

NAVAL POSTGRADUATE SCHOOL MONTEREY, CALIFORNIA



THESIS

**NUMERICAL INVESTIGATION OF SUBSONIC FLOW
OVER BODIES OF REVOLUTION AND A TYPICAL
MISSILE FOREBODY**

by

Ronald E. Marvin

September, 1995

Thesis Advisor:

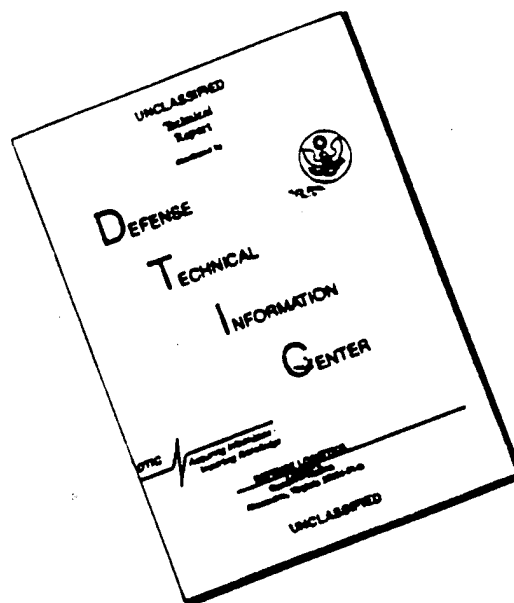
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Ronald E. Marvin
B. S., United States Military Academy, 1966

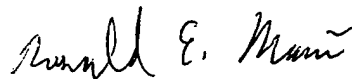
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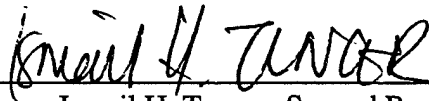


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ABSTRACT

It was the objective of this investigation to study the subsonic flow over a missile forebody. To this end the NASA Ames developed OVERFLOW, three dimensional, Navier-Stokes Code was applied and detailed results were obtained for the missile forebody. The flow conditions were set at a Mach Number of 0.3 and Reynolds Number of two million for each angle of attack at 0, 2, 6, 10 and 14 degrees. These viscous flow solutions were compared with three inviscid flow solutions, namely Slender Body Theory, von Karman's Method and the NASA Ames Panel Code called PMARC. As expected, the OVERFLOW and PMARC solutions were in good agreement at zero and small angles of attack. As the angle of attack was increased, the OVERFLOW solution showed the development of the vortical flow separation on the leeward side of the body.

TABLE OF CONTENTS

I.	INTRODUCTION	1
II.	SLENDER BODY METHOD FOR APPROXIMATING FLOW PAST BODIES OF REVOLUTION AT ZERO DEGREE ANGLE OF ATTACK ...	3
A.	THEORY	3
B.	COMPUTER IMPLEMENTATION OF THE METHOD.....	4
C.	MAPLE PROCEDURE VERIFICATION	5
D.	SUMMARY	7
II.	VON KARMAN'S METHOD FOR APPROXIMATING FLOW PAST BODIES OF REVOLUTION.....	9
A.	THEORY	9
B.	COMPUTER IMPLEMENTATION OF THE METHOD.....	11
C.	ERROR NORM ANALYSIS.....	12
D.	SUMMARY	14
IV.	PANEL METHOD FOR APPROXIMATING FLOW PAST BODIES OF REVOLUTION.....	15
A.	THEORY	15
B.	MODELING WAKES FROM BODIES OF REVOLUTION.....	17
C.	VERIFICATION OF PMARC SPHERICAL BODY MODEL	18
D.	VERIFICATION OF PMARC SPINDLE BODY MODEL AT ZERO DEGREE ANGLE OF ATTACK	19
E.	VERIFICATION OF PMARC SPINDLE BODY MODEL AT FIVE DEGREE ANGLE OF ATTACK	21
F.	SUMMARY	23
V.	A CFD FLOW MODEL OVER A TYPICAL MISSILE FOREBODY	25
A.	A BRIEF THEORY FOR THE OVERFLOW CODE	25
1.	Description of Basic Fluid Dynamic Equations	25
2.	Description of the Major Mathematical Operations	26
B.	OVERFLOW MISSILE FOREBODY MODEL DESCRIPTION	28

C.	OVERFLOW MISSILE FOREBODY FLOW SOLUTION CONVERGENCE STUDY	29
D.	PMARC MISSILE FOREBODY MODEL DESCRIPTION	29
E.	COMPUTING TIMES FOR OVERFLOW AND PMARC	34
F.	OVERFLOW AND PMARC C_p DISTRIBUTION COMPARISONS AT VARIOUS ANGLES OF ATTACK	34
G.	SUMMARY	47
VI.	CONCLUSIONS AND RECOMMENDATIONS	49
APPENDIX A. MAPLE PROCEDURE TO CALCULATE C_p VALUES ON A SPINDLE		51
APPENDIX B. FORTRAN LISTING OF THE SPHERE CODE		61
APPENDIX C. SPH.OUT FILE FROM THE SPHERE CODE		63
APPENDIX D. ERR.OUT FILE FROM THE SPHERE CODE		71
APPENDIX E. PMARC INPUT FILE FOR A SPHERE		73
APPENDIX F. PMARC INPUT FILE FOR A SPINDLE		75
APPENDIX G. OVERFLOW INPUT FILE FOR A MISSILE FOREBODY		77
APPENDIX H. PMARC INPUT FILE FOR A MISSILE FOREBODY		79
LIST OF REFERENCES		115
INITIAL DISTRIBUTION LIST		117

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I. INTRODUCTION

Over the years, a number of methods have been developed to provide the missile aerodynamicist with tools to predict the aerodynamic characteristics of various missile configurations. They range from purely empirical methods and simple slender body theory to panel methods and Navier-Stokes Methods. A recent review of these methods can be found in Reference 1. This review reveals the fact, that the prediction of strong viscous and separated flow effects, presents a great challenge.

Modern missiles tend to have small lifting surfaces. Therefore, the contributions of the body to the forces and moments, and the understanding of the wing-body interference becomes very important. In addition, the missiles are launched at relatively high angles of attack, further exacerbating the problem, due to the vortex shedding from the missile body. As a result, there is a growing need to compute the detailed flow separation and the vortical flow features, which occur over typical modern missile configurations, using the full viscous flow equations.

At NASA-Ames a considerable amount of work was done to study the viscous flow surfaces over blunt-nose bodies of revolution. This work used the NASA-Ames developed OVERFLOW Code for the solution of the thin-layer Navier-Stokes equations. References 2 through 5 document this effort. Recently Reference 6 extended this work, by computing of the flow field about a complete missile at high angle of attack. The results of Reference 6 were compared to experiments, described in Reference 7.

As pointed out, for example in Reference 8, vortices are shed from the forward body and the canards. These vortices affect the pressure distribution and the forces and moments on the missile. Therefore, in this thesis the missile forebody of Reference 6 is investigated in greater detail.

Starting with the solution for zero angle of attack, the angle of attack is gradually increased to 14 degrees. This allows the comparison of the Navier-Stokes solutions with the simpler and faster methods for the more benign flows at small angles of attack, where inviscid methods can be expected to give reasonable results. This comparison allows assessing the range of validity of the simpler methods. Slender Body Theory, von Karman's Method and NASA-Ames panel code PMARC were chosen to compare with the Navier-Stokes calculations from the OVERFLOW Code. These methods are briefly described in Chapters 2 through 4. Chapter 5 presents the major results of this thesis, while Chapter 6 gives the conclusions and recommendations for future work.

II. SLENDER BODY METHOD FOR APPROXIMATING FLOW PAST BODIES OF REVOLUTION AT ZERO DEGREE ANGLE OF ATTACK

A. THEORY

The Slender Body Method is based on the assumptions of incompressible, inviscid and irrotational flow. The governing equation is Laplace's equation for the potential $\phi(x, r, \theta)$ [Reference 9],

$$\frac{\partial^2 \phi}{\partial x^2} + \frac{\partial^2 \phi}{\partial r^2} + \frac{1}{r} \frac{\partial \phi}{\partial r} + \frac{1}{r^2} \frac{\partial^2 \phi}{\partial \theta^2} = 0 \quad (2.1)$$

Equation 2.1 is in cylindrical coordinates (x, r, θ) . The body's angle of attack is assumed to be zero, and the body's longitudinal axis is assumed to coincide the x axis.

Source superposition along the x axis, then produces the well-known equation for the disturbance potential,

$$\phi = -\frac{U}{4\pi} \int_0^l \frac{F'(\xi) d\xi}{[r^2 + (\xi - x)^2]^{3/2}} \quad (2.2)$$

(U) is the free stream speed. (ξ) is the source location on the body's longitudinal axis. The body extends from 0 to l . (F') is the change of the body's cross sectional area along the longitudinal axis. It is given by,

$$F'(x) = 2\pi R(x) \frac{dR(x)}{dx} \quad (2.3)$$

The body radius is given by,

$$r = R(x) \quad (2.4)$$

Equation 2.2 is obtained by imposing the slender body assumption, which requires that the body's maximum diameter is much less than the body's length. Also the absolute magnitude of the change in the body's radius $\left(\frac{dR(x)}{dx}\right)$ must be small.

From Reference 10. Equation 2.2 can be rewritten in non-dimensional form, obtaining the non-dimensional longitudinal velocity component,

$$\frac{u_x - U}{U} = -\frac{1}{4\pi} \int_0^1 \frac{F'(\xi)(\xi - x)d\xi}{[r^2 + (\xi - x)^2]^{3/2}} \quad (2.5)$$

The non-dimensional radial, velocity component becomes,

$$\frac{u_r}{U} = \frac{r}{4\pi} \int_0^1 \frac{F'(\xi)d\xi}{[r^2 + (\xi - x)^2]^{3/2}} \quad (2.6)$$

Using these components, the coefficient of pressure (C_p) becomes,

$$C_p = -2\left(\frac{u_x - U}{U}\right) - \left[\left(\frac{u_r}{U}\right)^2 + \left(\frac{u_x - U}{U}\right)^2\right] \quad (2.7)$$

Depending on the complexity of the surface function, the integrals in Equations 2.5 and 2.6 can be difficult to integrate analytically. Therefore, the MAPLE Program was used to symbolically integrate Equations 2.5 and 2.6 and solve Equation 2.7.

B. COMPUTER IMPLEMENTATION OF THE METHOD

MAPLE is a general purpose, commercially available, computer code that can perform symbolic mathematics [Reference 11]. The code may also be used for numerical mathematics. To demonstrate this program's usefulness in potential flow theory, Appendix D lists the implementation of Equations 2.5, 2.6 and 2.7 in Sections 1 through 3 respectively. The MAPLE Procedure, given in these Sections, models the flow over a spindle body of revolution as shown in Figure 1. The spindle is at zero degree angle of attack.

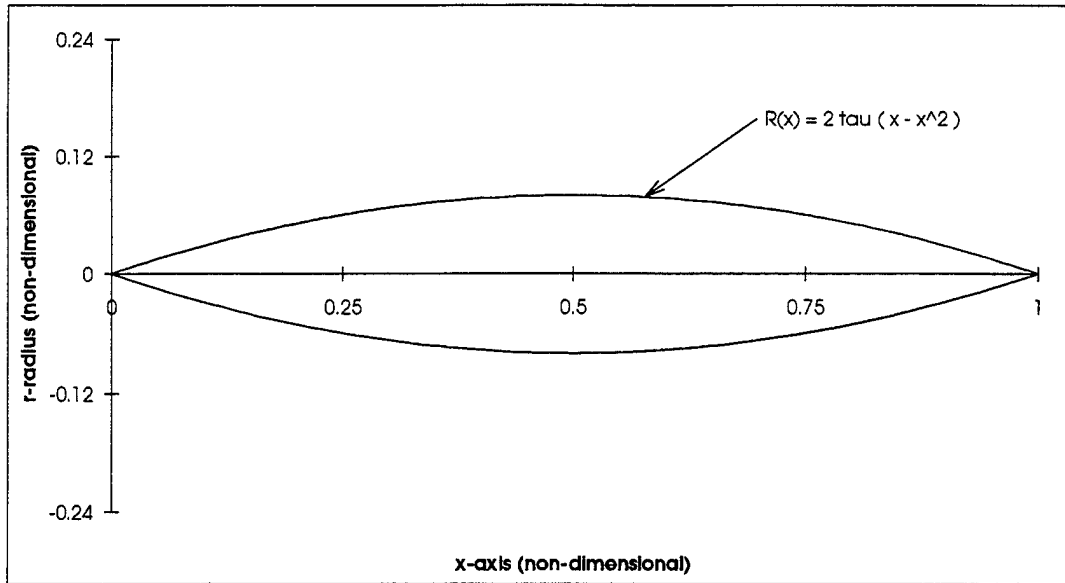


Figure 1. Spindle Body of Revolution Surface Function and Shape.

The geometry is non-dimensional, so that the length is one and maximum diameter (τ) is 16% of the body's length. The maximum diameter occurs at the middle of the body. Velocity components and C_p values for 10 points on the spindle's surface are given in Section 3 of Appendix A.

C. MAPLE PROCEDURE VERIFICATION

The parabolic spindle shown in Figure 1 is described by,

$$R(x) = 2\tau(x - x^2) \quad (2.8)$$

Its first derivative becomes,

$$\frac{dR(x)}{dx} = 2\tau(1 - 2x) \quad (2.9)$$

The absolute magnitude of Equation 2.9 will always be less than one, since the maximum diameter is 0.16. This satisfies the slender body approximation.

To verify the spindle results in Section 3 of Appendix A, Reference 10 gives an approximation of Equation 2.5 of the form,

$$\frac{u_x - U}{U} = \frac{1}{2\pi}(F'')\ln r - \frac{1}{2\pi}\left\{\left(F'\right)\frac{\left(\frac{1}{2}-x\right)}{x(1-x)} + (F'')\left[\ln 2\sqrt{x(1-x)} - 1\right] + \right.$$

$$\left. \frac{1}{2}(F''')\left(\frac{1}{2}-x\right) + \frac{1}{48}(F^{iv})\left[1 + 4\left(\frac{1}{2}-x\right)^2\right]\right\} \quad (2.10)$$

In this equation (F') is Equation 2.3, while (F''), (F''') and (F^{iv}) represent second, third and fourth derivatives of the body's cross sectional area respectively. Section 4 of Appendix A determines velocity values for Equation 2.10. These are compared to values derived from Equation 2.5, using the surface function for the spindle in Figure 1. Figure 2 shows the comparison between Equations 2.5 and 2.10 for 48 surface points. The agreement is seen to be excellent. Both solutions exhibit the well-known singular behavior of slender body theory at the nose and tail of the body.

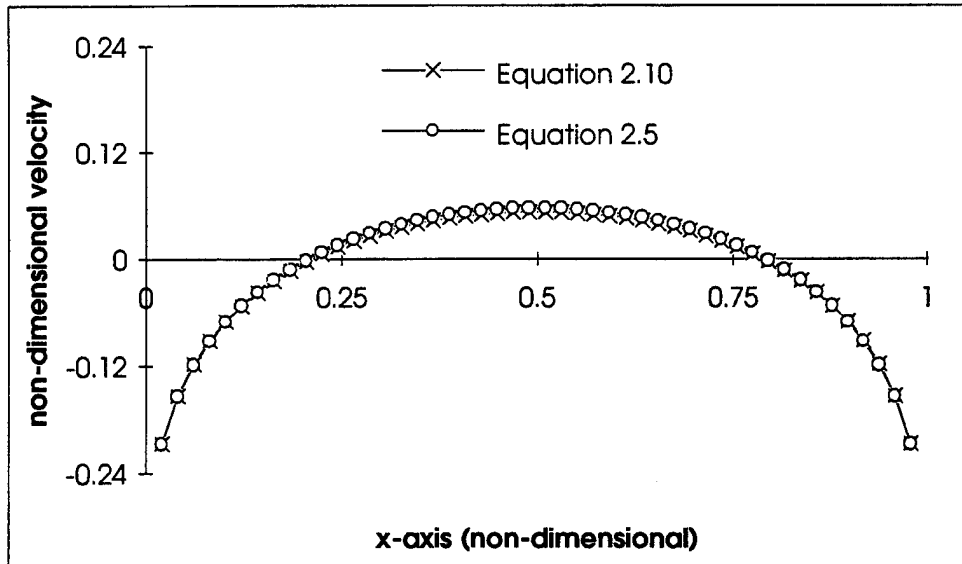


Figure 2. Comparison of Equation 2.5 and 2.10.

D. SUMMARY

This singularity behavior is the result of the integration of Equations 2.5 and 2.6 in Sections 1 and 2 of Appendix A respectively. In both cases integration yields terms containing the logarithmic expression,

$$\ln(2\sqrt{x^2 + r^2} - 2x) \quad (2.11)$$

At the spindle's leading or trailing tip the radius approaches zero. At the tips the x coordinate approaches either zero or one respectively. Under these conditions the logarithmic expression becomes indeterminate. As depicted in Figure 2, the velocity components do not have a finite value at the tips of the body. Expression 2.11 also appears in Equation 2.7 (see Section 3 of Appendix A), indicating that the C_p at the tips will not be the expected finite value of one. Another indication of this indeterminate behavior is given in Sections 5 and 6 of Appendix A.

In Section 5 as r and x approach zero, the C_p value peaks at approximately 0.89. The values decrease and do not approach the expected C_p value of one. In Section 6 as r approach's zero and x approach's one, the C_p value oscillates about 0.64, then suddenly decreases at the x coordinate of 0.99927. At this point the MAPLE Program reports a singularity in one of the logarithmic expressions occurring in the solution of Equation 2.7.

This implies that the Slender Body Method is inherently inaccurate near the leading and trailing tips of a body of revolution. The method can be used to confirm computational fluid dynamic (CFD) models in the middle portion of the body, provided that the surface function is relatively simple and continuous. When surface functions can not be easily formulated for complex shapes, the Panel Method might prove more useful.

III. VON KARMAN'S METHOD FOR APPROXIMATING FLOW PAST BODIES OF REVOLUTION

A. THEORY

von Karman's Method assumes a potential flow about a body of revolution [Reference 12]. The flow is considered incompressible, inviscid and irrotational. The governing equation is Laplace's equation,

$$\frac{\partial^2 \psi}{\partial z^2} + \frac{\partial^2 \psi}{\partial r^2} + \frac{1}{r} \frac{\partial \psi}{\partial r} + \frac{1}{r^2} \frac{\partial^2 \psi}{\partial \theta^2} = 0 \quad (3.1)$$

This represents the flow within a body of revolution in cylindrical coordinates (z, r, θ) [Reference 13]. The solution to this equation may be formed by superposition of elementary flows. These are represented by stream functions (ψ). The stream function for uniform flow of speed (U) z direction is given by,

$$\psi = \frac{1}{2} U r^2 \quad (3.2)$$

To obtain a solution for flow over a body of revolution at zero angle of attack (non-lifting flow), whose axis coincides with the z axis, it is sufficient to distribute sources along the body axis. The stream function of a source located in the origin of the coordinate system is,

$$\psi = -\frac{mz}{\sqrt{r^2 + z^2}} \quad (3.3)$$

The change in stream function ($d\psi$) at any point in the flow from sources along a small interval ($d\zeta$) there is,

$$d\psi = -\frac{q_j(z-\zeta)d\zeta}{\sqrt{r^2 + (z-\zeta)^2}} \quad (3.4)$$

The interval is measured along the body's longitudinal axis. (q_j) is the source strength per unit length. Integration of Equation 3.4 along this axis yields the induced stream function at any point in the flow. This function is created by sources on the longitudinal axis. The effect of all the source strengths and uniform flow are summed to give the total stream function at any point in the flow. From Reference 13 the summation can be represented as,

$$\psi(r,z) = \frac{1}{2}Ur^2 + \sum_j \left[\sqrt{r^2 + (z - z'_{j+1})^2} - \sqrt{r^2 + (z - z'_j)^2} \right] q_j \quad (3.5)$$

The (z'_{j+1}) and (z'_j) coordinates define the location of the source of constant strength (q_j) on the longitudinal axis. The body of revolution is defined by Equation 2.4. Realizing that the body surface represents the stagnation streamline $\psi = 0$, equating the right hand side of Equation 3.5 to zero provides a means to determine the unknown source strength q_j .

For finite length bodies, the total sum of the source strengths must be zero. This additional condition must be appended to the system given by Equation 3.5. The system therefore becomes,

$$\{\mathbf{const}\} = [\mathbf{dd}]\{\mathbf{q}\} \quad (3.6)$$

The vector $\{\mathbf{const}\}$ is only a function of the known body radius. The coefficient matrix $[\mathbf{dd}]$ is a function of the known body geometry and source locations, while the vector $\{\mathbf{q}\}$ is the unknown source strengths. The resulting system of algebraic equations may now be solved by common equation solves, such as Cramer's Rule, Gaussian Elimination, etc.

Once the source strengths are known on the body's surface, the flow velocity components (u_r, u_z) can be defined through the first derivative of Equation 3.5. From Reference 13 the first derivative with respect to z yields the radial component u_r . The longitudinal component u_z is obtained from the first derivative with respect to r . Using Bernoulli's Equation, the non-dimensional coefficient of pressure (C_p) is obtained as,

$$C_p = 1 - \left(\frac{u_r}{U} \right)^2 - \left(\frac{u_z}{U} \right)^2 \quad (3.7)$$

The C_p is the ultimate result of this Method. Since the C_p is a function of the body's geometry, the value will differ at various points in the outer flow field and on the body's

surface. This condition makes the C_p parameter an excellent metric to make comparisons of results from various analytic and computer code techniques.

B. COMPUTER IMPLEMENTATION OF THE METHOD

Reference 13 lists a computer code that implements von Karman's Method. The code was called SPHERE. A FORTRAN listing of the code is given in Appendix B. This code predicts C_p values about a sphere of radius (a). Since the flow field is symmetrical, the code only calculates C_p values at surface points on the upper half of the sphere. The free stream velocity and the radius of the sphere were assumed to be one. SPHERE used Cramer's Rule to solve the system of equations.

Appendix C and D contain listings of the two output files created by SPHERE. The file called SPH.OUT in Appendix C gives the surface point location (r_i, z_i), calculated (C_p) and exact C_p (C_{pex}) for various numbers of surface points on the sphere. The exact C_p values are known to be given by,

$$C_{pex} = 1 - \frac{9}{4} \left(\frac{r_i}{a} \right)^2 \quad (3.8)$$

The derivation of Equation 3.8 may be found in Reference 13. Calculated and exact C_p values were determined for nine surface points as a check of the code. Figure 3 shows that the calculated C_p values plot identically on the exact C_p values for the nine points. However, when the number of surface points is increased, the calculated C_p values diverge from the exact. This is expected because the matrix becomes increasingly ill-conditioned.

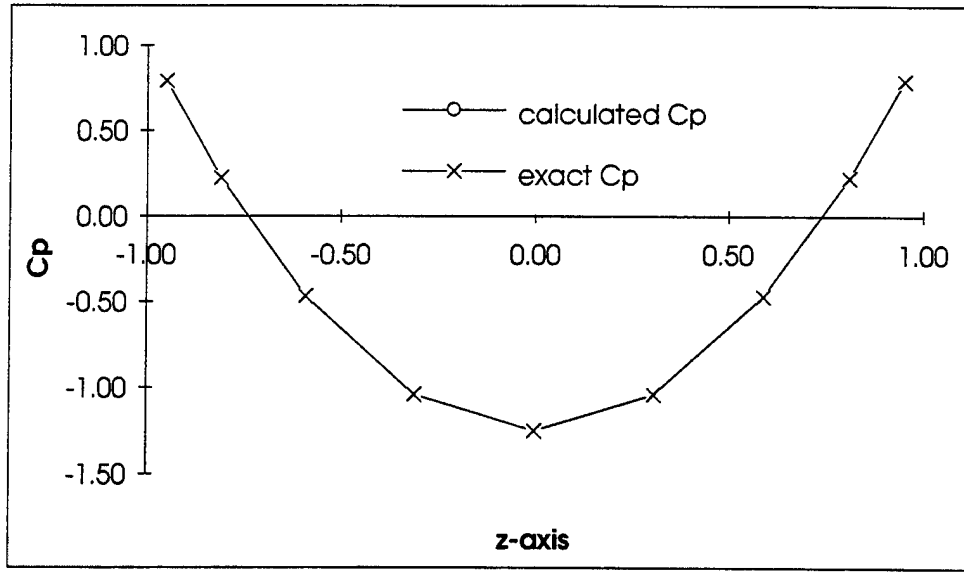


Figure 3. Calculated and Exact C_p Distributions for 9 Surface Points on a Sphere.

C. ERROR NORM ANALYSIS

Figures 4 and 5 show how the calculated C_p distributions diverge from the exact distribution, as the number of surface points increases. This divergent behavior comes from the ill-conditioning of the coefficient matrix $[dd]$ in Equation 2.6. The growth of this ill-conditioning can be defined as,

$$\{\mathbf{error}\} = [dd]\{\mathbf{q}\} - \{\mathbf{const}\} \quad (3.9)$$

$$error \text{ norm} = \sqrt{\sum \{\mathbf{error}\}^2} \quad (3.10)$$

Appendix D lists the ERR.OUT file from the SPHERE Code. This listing gives the results of Equation 3.10 for various numbers of surface points. Examination of this data and the data's plot in Figure 6 clearly shows the error growth, as the number of surface points increases. When the number of surface points exceeds 39, the matrix $[dd]$ becomes so ill-conditioned that the code cannot obtain the vector $\{\mathbf{q}\}$ in Equation 3.6.

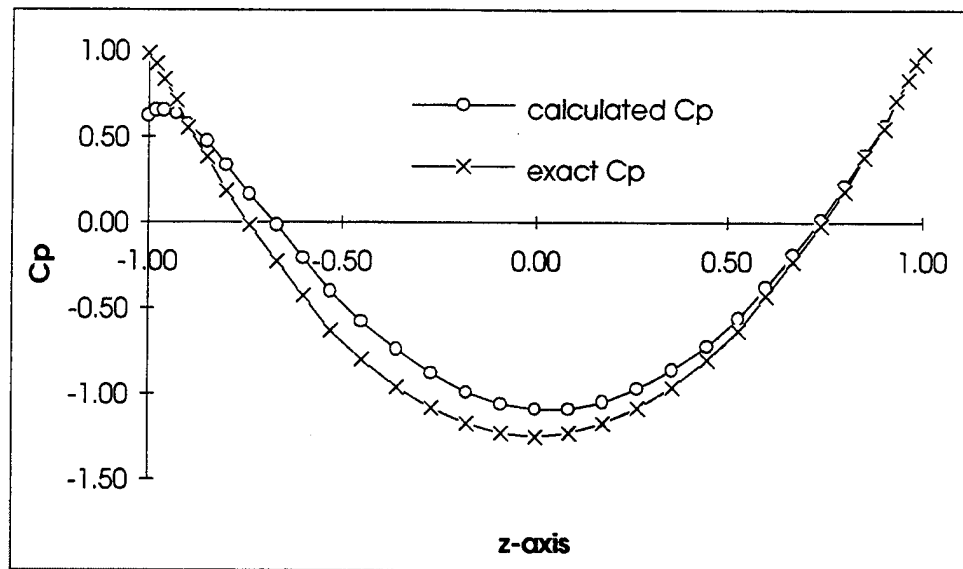


Figure 4. Calculated and Exact C_p Distributions for 33 Surface Points on a Sphere.

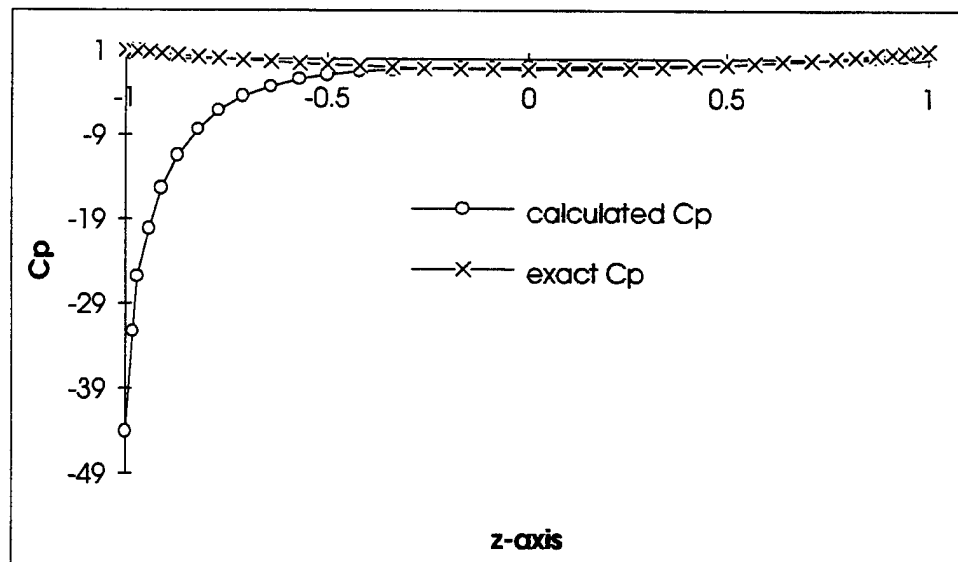


Figure 5. Calculated and Exact C_p Distributions for 35 Surface Points on a Sphere.

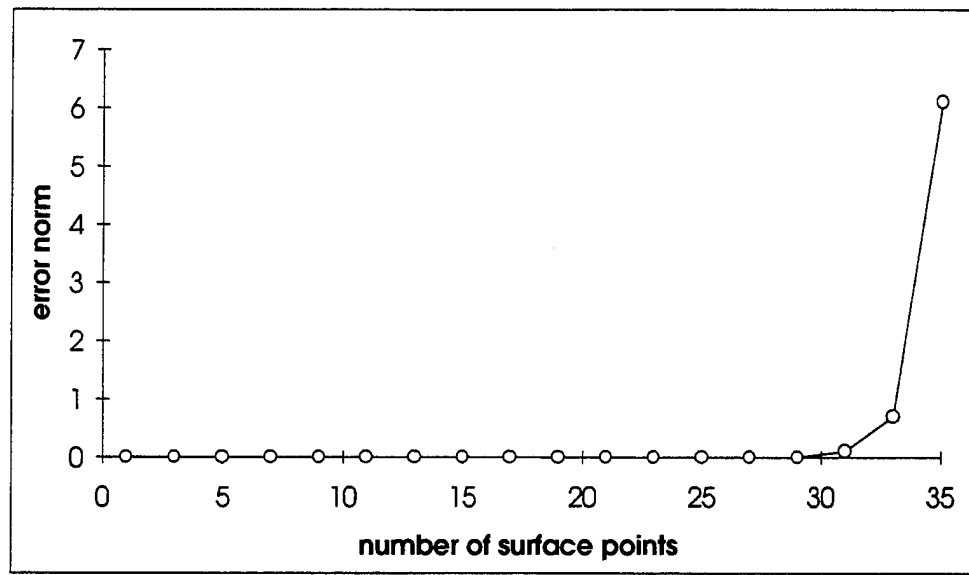


Figure 6. Error Norm versus Number of Surface Points for a Spherical Body of Revolution.

D. SUMMARY

As shown by von Karman, this method of representing incompressible, inviscid flow over a sphere can be extended to the flow over arbitrary bodies of revolution at zero and small angles of attack. However, the method is limited to slender bodies, because source or doublet distributions are used only along the axis. In contrast to the slender body theory described in Chapter II., it has the advantage of describing the flow over blunt-nosed bodies.

A more general method that eliminates the restriction to slender bodies is afforded by surface distributions of sources and doublets. Such a method is described in the next chapter.

IV. PANEL METHOD FOR APPROXIMATING FLOW PAST BODIES OF REVOLUTION

A. THEORY

The Panel Method is also based on potential flow theory and generally parallels the development of the Slender Body and von Karman Methods. Reference 14 gives a detailed mathematical description of the Panel Method. This reference also describes how the Method is implemented in the computer code called PMARC (Panel Method Ames Research Center). PMARC assumes an incompressible and irrotational flow. The code does allow a viscous boundary layer on the body, but this option was not used. All of the PMARC results in this thesis were based on inviscid flows. The method solves the three dimensional form of Laplace's equation,

$$\nabla^2 \phi = 0 \quad (4.1)$$

As shown in Reference 14, application of Green's Theorem leads to the following integral equation,

$$0 = \left[\iint_{S-P} \mu \bar{n} \cdot \nabla \left(\frac{1}{\bar{r}} \right) dS - 2\pi \mu_P \right] + \iint_S \left(\frac{\sigma}{\bar{r}} \right) + \iint_W \mu_w \bar{n} \cdot \nabla \left(\frac{1}{\bar{r}} \right) dS \quad (4.2)$$

(μ) and (μ_w) are unknown doublet strengths, on the body's surface element (dS) and on the wake (W) respectively. The doublet strength (μ_P) is zero, if the arbitrary point (P) of interest is off the surface. When the point is on the surface, the strength is 2π . (\bar{n}) is the outward surface normal on the body's surface, while (\bar{r}) is the vector between P and dS . \bar{n} and \bar{r} are known from the body's geometry. The source strength (σ) is given by,

$$\sigma = \frac{1}{4\pi} (V_{norm} - \bar{n} \cdot \bar{V}_\infty) \quad (4.3)$$

On the body's surface the normal velocity (V_{norm}) is usually zero or may be prescribed. The free stream velocity vector (\bar{V}_∞) is known. Therefore, the source strength is known in Equation 4.2.

Equation 4.2 may be solved numerically by discretizing the body's surface into panels. These panels represent areas of source and doublet strengths, that when known, will define the flow field over the body's surface. This marks the only difference between the Panel Method and those given in Chapters II and III. Previous methods located source strengths inside the body on its' longitudinal axis. In PMARC the source and doublet strengths are assumed to be constant over a panel.

The discretization process results in a system of algebraic equations of the form,

$$\sum_{k=1}^{N_s} (\mu_k C_{jk}) + \sum_{k=1}^{N_s} (\sigma_k B_{jk}) + \sum_{l=1}^{N_w} (\mu_{w_l} C_{jl}) = o|_{j=1, N_s} \quad (4.4)$$

The summation terms in Equation 4.4 are analogous to the integrals of Equation 4.2. (B) and (C) become respectively,

$$B_{jk} = \iint_k \frac{1}{\bar{r}} dS \quad (4.5)$$

$$C_{jk} = \iint_k \bar{n} \cdot \nabla \left(\frac{1}{\bar{r}} \right) dS, \quad C_{jj} = -2\pi \quad (4.6)$$

The first and second terms in Equation 4.4 represent the placement of surface panels on the body. The third term accounts for the wake panels trailing downstream from the body. In PMARC this equation is solved by an iterative technique for very large matrices. The solution occurs in a time stepping fashion, so that either steady or unsteady flows may be calculated. The wake doublet strengths become a function of the surface doublet strengths or,

$$\mu_w = \mu_w(\mu) \quad (4.7)$$

This means that the PMARC time stepping algorithm effectively substitutes Equation 4.7 into Equation 4.4 to yield the surface doublet strengths directly.

These doublet strengths are differenced to obtain the tangential velocity components on the surface of each panel. The velocity components are then used in,

$$C_{p_k} = 1 - \frac{\bar{V}_k^2}{\bar{V}_\infty^2} + \left(\frac{8\pi}{\bar{V}_\infty^2} \right) \left(\frac{\mu(t) - \mu(t-1)}{\Delta t} \right) \quad (4.8)$$

Equation 4.8 defines the pressure coefficient (C_{p_k}) at each panel (k). (\bar{V}_k) is the flow velocity magnitude at each panel. The third term in Equation 4.8 accounts for pressure changes when the flow is unsteady. This term is zero for steady flows, so that Equation 4.8 becomes identical to Equations 2.7 and 3.7. Because the theory and PMARC allow the effects of wakes, a problem arises as to how to attach wake panels to bodies of revolution to simulate the effect of flow separation.

B. MODELING WAKES FROM BODIES OF REVOLUTION

PMARC allows wakes to be excluded from the model. When this occurs PMARC will calculate C_p values, but can not calculate circulation about the body. In this case total aerodynamic forces, such as lift, drag, etc. are not determined. Alternatively, when wakes are included in the model, they must be defined as either *flexible* or *rigid*. Flexible wakes require that the separation line be specified in the input file. Rigid wakes require initial wake panels be defined in regions where the flow is expected to separate.

Reference 14 strongly implies that PMARC's wakes should only be modeled when attached to traditionally defined lifting surfaces, that have sharp, linear trailing edges. This allows a precise definition of a separation line or initial wake panel layout along the edge. The code may then properly apply the Kutta condition at the edge. When wakes are attached to bodies of revolution, without lifting surfaces, the results appear to be questionable, depending on whether the body has a tapered, hemispherical or flat base [Reference 15]. A literature search of References 16 through 19 provided no guidance into acceptable wake modeling for bodies of revolution. Reference 18 claimed to have modeled wakes on a body with no lifting surfaces, but provided no details of the PMARC input file.

In view of this situation the author did not include wakes in any of the PMARC flow models in the thesis. All flow models were considered to be steady. This meant that the third terms of both Equations 4.4 and 4.8 had a value of zero. To evaluate PMARC's ability to compute fully attached flows the code was first applied to the sphere and the spindle, so that the results could be compared with the Slender Body and von Karman's Methods.

C. VERIFICATION OF PMARC SPHERICAL BODY MODEL

A spherical body was constructed for the PMARC Code. Appendix E lists the input file for the sphere. In the file the name lists &BINP8, &BINP8A and &BINP8B were set to zero in the file. These settings created steady flow. The parameters in the &WAKE1 command were also set to zero. This eliminated the wakes from the model. Figure 7 shows the surface paneling layout.

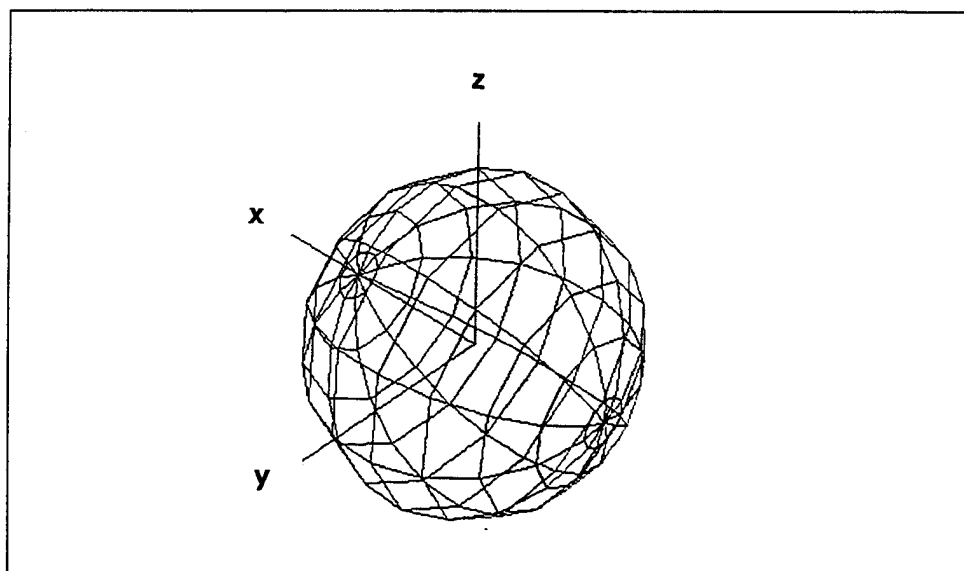


Figure 7. Surface Panel Layout for a Spherical Body.

The panels were positioned on the sphere to yield nine C_p values that corresponded to identical radial and longitudinal locations of the exact C_p values given in Reference 13. All PMARC results shown in this thesis were generated by version 12.21, 4 March 1994, of the code. Figure 8 shows excellent comparison between PMARC and exact C_p values.

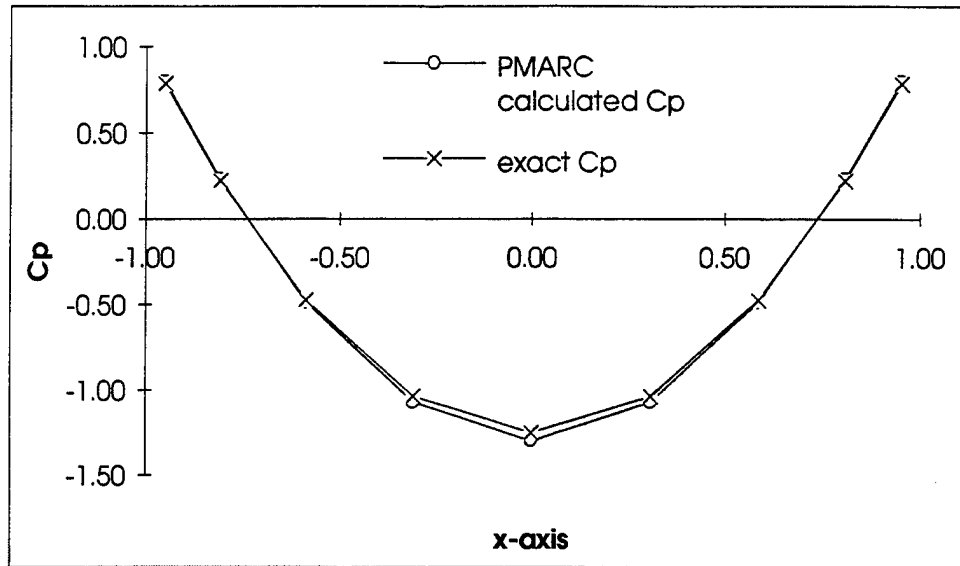


Figure 8. PMARC and Exact C_p Distributions for nine Surface Points on a Sphere without wakes.

D. VERIFICATION OF PMARC SPINDLE BODY MODEL AT ZERO DEGREE ANGLE OF ATTACK

This trend was also confirmed when the spindle body was modeled in steady flow, without wake panels and at zero degree angle of attack. A typical PMARC input file for the spindle is given in Appendix F. Figure 9 displays the PMARC spindle model, using the surface function given in Figure 1. In Figure 9 a positive rotation about the y-axis produces a nose up positive angle of attack, using the right hand rule. The geometry was non-dimensionalized similar to Figure 1. C_p values were calculated and compared to similar values from the Slender Body Method, described in Chapter II. Figure 10 shows that PMARC results compare favorably.

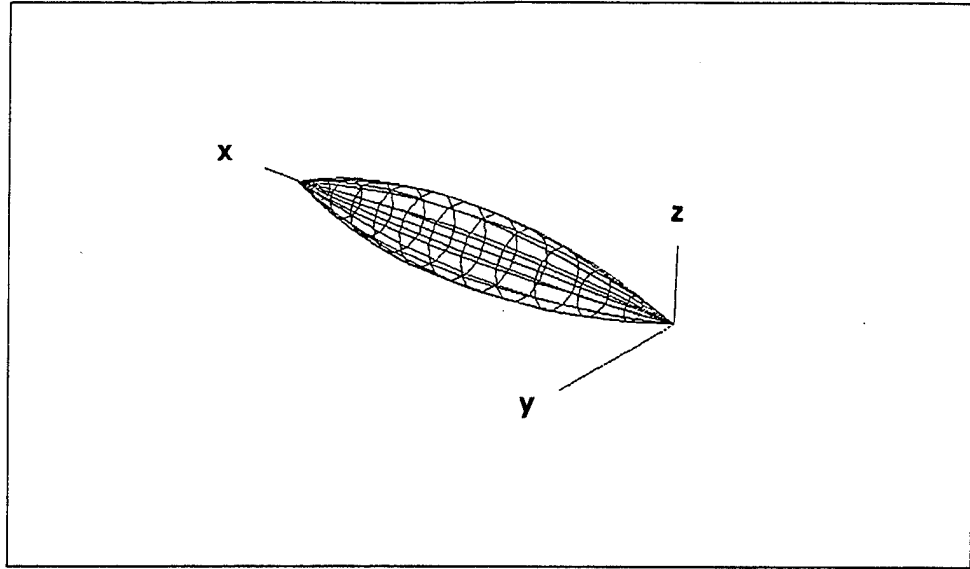


Figure 9. Surface Panel Layout for a Spindle Body.

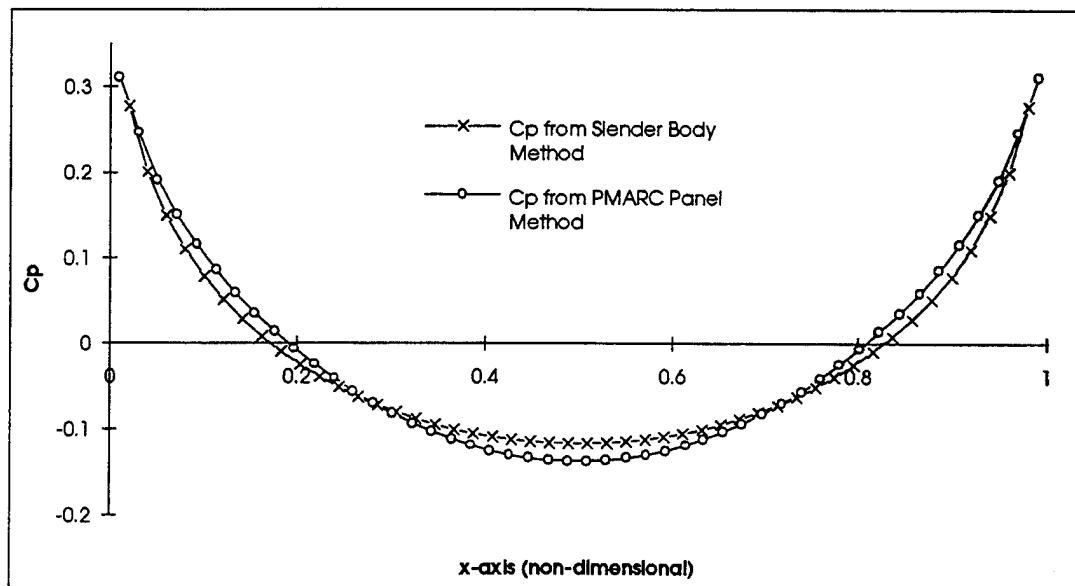


Figure 10. PMARC and Slender Body C_p Distributions for 48 Surface Points on a Spindle at Zero Degree Angle of Attack without Wakes.

E. VERIFICATION OF PMARC SPINDLE BODY MODEL AT FIVE DEGREE ANGLE OF ATTACK

The PMARC spindle model was exercised at both zero and five degree angles of attack. This was done to verify the model's physical behavior from a C_p point of view. Figure 11, 12 and 13 compare C_p distributions on longitudinal lines along the body's suction or *top* surface, middle or *mid plane* surface and pressure or *bottom* surface respectively. These longitudinal lines can, alternatively, be defined in Figure 9 as lying in the positive x, z plane for the top surface, the positive x, y plane for the mid plane surface and the positive y, negative z plane for the bottom surface. As expected for a symmetrical spindle, at zero degree angle of attack the C_p distributions are the same longitudinally, regardless of surface orientation.

When the spindle is elevated to an attack angle of five degrees, suction or negative pressures develop on the forward top surface in Figure 11. As expected, the flow accelerates over the forward top surface, creating negative pressures. On the aft top surface the flow experiences an adverse pressure gradient, slowing the flow and creating positive pressure.

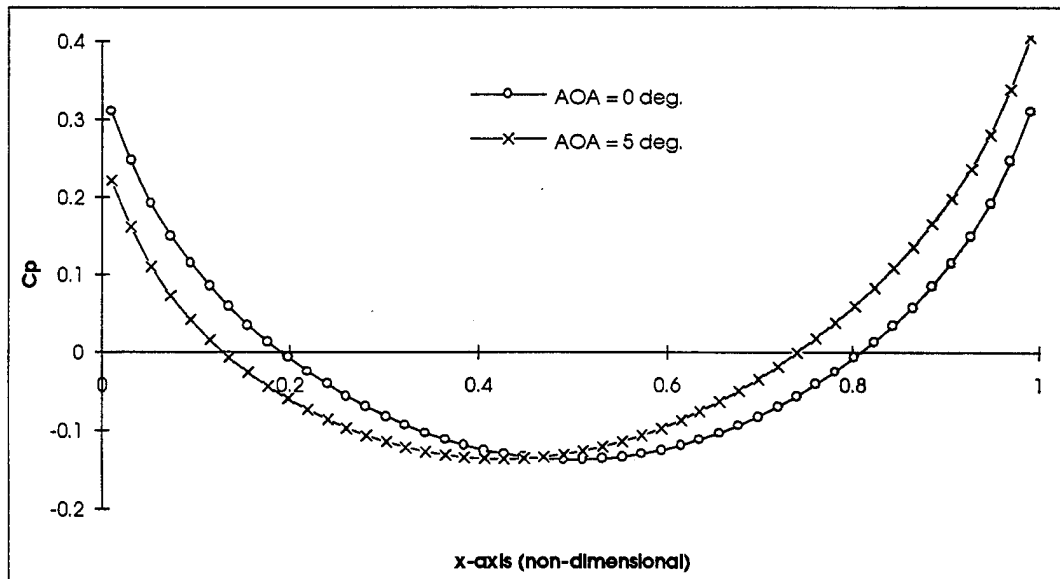


Figure 11. PMARC C_p Distributions for 48 Surface Points on a Spindle at Zero and Five Degrees' Angles of Attack on the Top Surface without Wakes.

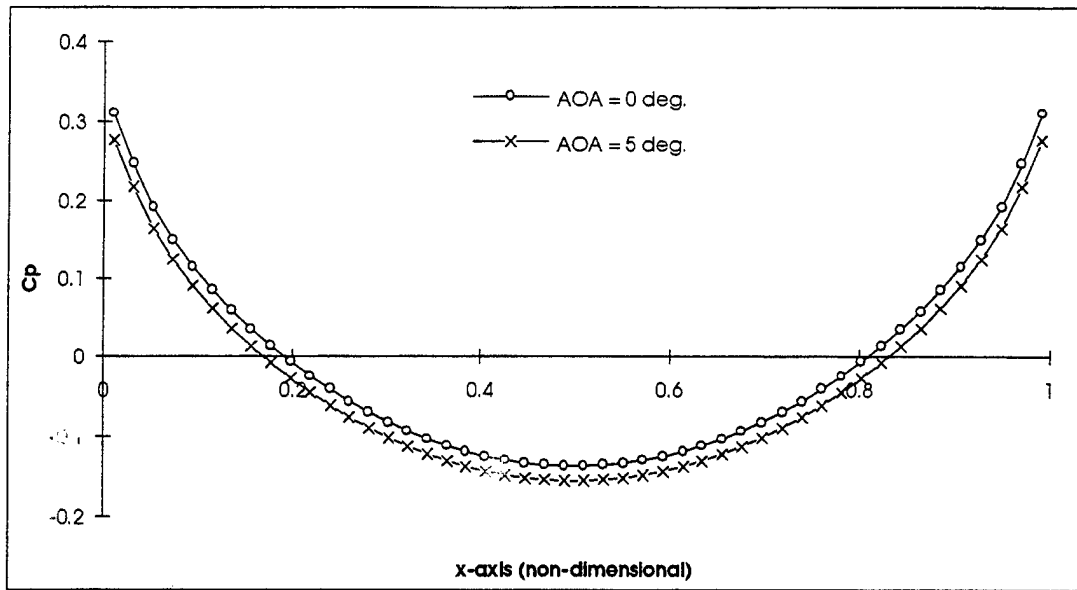


Figure 12. PMARC C_p Distributions for 48 Surface Points on a Spindle at Zero and Five Degrees' Angles of Attack on the Mid Plane Surface without Wakes.

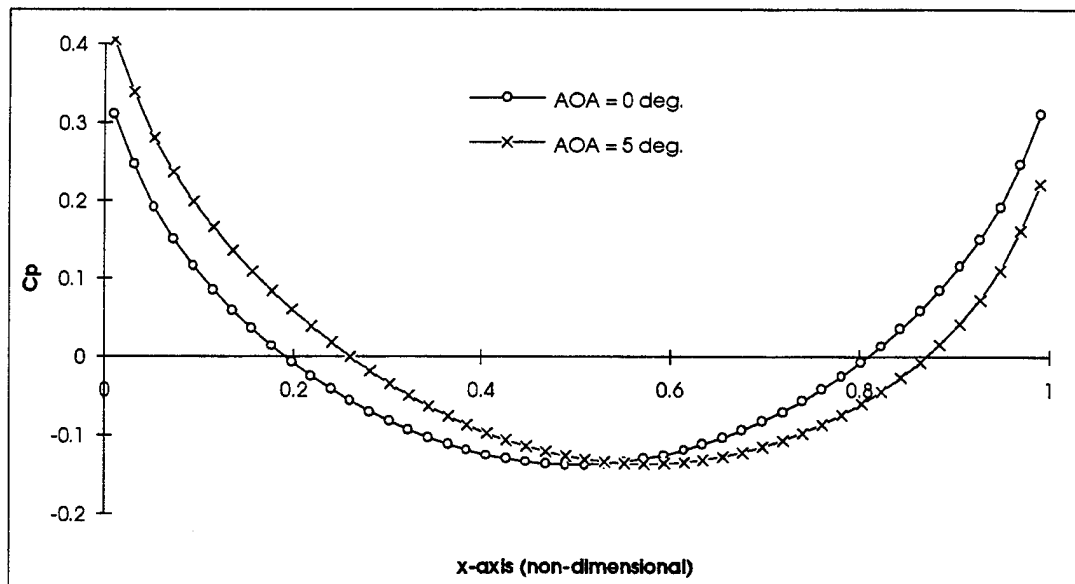


Figure 13. PMARC C_p Distributions for 48 Surface Points on a Spindle at Zero and Five Degrees' Angles of Attack on the Bottom Surface without Wakes.

Figure 12 indicates a relatively constant difference between C_p values at attack angles of zero and five degrees along the spindle's mid plane surface. At zero degree a fluid particle experiences only body fixed, x and y velocity components, as it travels along the body's mid plane surface [see Figure 9]. The particle travels circumferentially and longitudinally along the body's side, when the body is rotated to five degrees. In this case the particle's velocity vector contains all three velocity components in the x, y and z directions. The total velocity is different for either case, producing different C_p distributions.

As expected, in Figure 13 the flow slows on the forward bottom surface, creating positive pressures. On the aft bottom surface the flow experiences a favorable pressure gradient, creating negative pressures.

F. SUMMARY

The Panel Method, as implemented in the PMARC Code, predicts C_p distributions for bodies of revolution that compare very well with classical potential theories. The code predicts the expected physical behavior of bodies at small angles of attack. Only steady, incompressible, inviscid and irrotational conditions should be considered in the flow. More investigation is needed to understand PMARC's predictions very near the tips of conical shaped bodies. Guide lines should be developed for modeling wakes on bodies of revolution.

Even with these slight deficiencies, PMARC is a suitable tool to confirm pressure fields about CFD models, as long as viscous effects are small. Before explaining how PMARC may be used to verify these models, a brief discussion of the CFD computer code used in this study will be given.

V. A CFD FLOW MODEL OVER A TYPICAL MISSILE FOREBODY

A. A BRIEF THEORY FOR THE OVERFLOW CODE

1. Description of Basic Fluid Dynamic Equations

Due to the experience discussed in Chapter I, the Navier-Stokes flow solver called OVERFLOW was chosen for this study. OVERFLOW is currently in development at the NASA-Ames Research Center.

From Reference 20 the governing equations may be written in the three Cartesian x , y and z coordinates. In the vector form the equation for conservation of the flow mass becomes,

$$\frac{\partial \rho}{\partial t} + \nabla \cdot \rho \vec{V} = 0 \quad (5.1)$$

The conservation of flow momentum in compact form is,

$$\rho \frac{D\vec{V}}{Dt} = \nabla \cdot \sigma_{ij} \quad (5.2)$$

Three momentum equations are obtained, when Equation 5.2 is expanded in each Cartesian direction. The conservation of energy equation in the flow may be written as,

$$\rho \frac{De_i}{Dt} = \nabla \cdot (k \nabla T) + \sigma_{ij} \frac{\partial u_i}{\partial x_j} \quad (5.3)$$

In these five equations (ρ) , (t) , (\vec{V}) , (e_i) , (k) and (T) are respectively density, time, the velocity vector, internal energy, thermal conductivity and temperature. The fluid stresses (σ_{ij}) are split into pressure and viscous terms as,

$$\sigma_{ij} = p\delta_{ij} + \mu \left(\frac{\partial u_i}{\partial x_j} + \frac{\partial u_j}{\partial x_i} \right) + \delta_{ij} \lambda \nabla \cdot \bar{V} \quad (5.4)$$

The first term in Equation 5.4 is the normal pressure (p) and (δ_{ij}) is the Kronecker Delta Function. This term represents the normal stress components on the fluid element. In the second term (μ) is the molecular viscosity and $\left(\frac{\partial u_i}{\partial x_j} + \frac{\partial u_j}{\partial x_i} \right)$ is the strain in the fluid. In the third term (λ) is the bulk viscosity and is usually assumed equal to $\left(-\frac{2}{3}\mu \right)$ [Reference 21]. When this assumption is made, the last two terms in Equation 5.4 may be combined to form the viscous or shearing stress components on the fluid element.

In OVERFLOW the molecular viscosity in the flow field is approximated by Sutherland's Formula as,

$$\frac{\mu}{\mu_o} \approx \left(\frac{T}{T_o} \right)^{3/2} \left(\frac{T_o + S}{T + S} \right) \quad (5.5)$$

(μ_o) and (T_o) are the reference flow viscosity and temperature respectively. The parameter (S) is assumed to be equal to 194 degrees Rankine for standard air. The turbulent viscosity is determined by the Baldwin and Lomax Turbulence Model in the boundary layer. This turbulent viscosity is added to Equation 5.5. Other turbulence models are available in the code [Reference 20].

Equations' 5.1, 5.2 and 5.3 allow modeling of compressible, viscous and vortical flows, removing the basic assumptions underlying the previous potential flow theory methods. Various mathematical operations are performed on Equations 5.1, 5.2 and 5.3. This effectively places them in a finite difference form suitable for a computer solution.

2. Description of the Major Mathematical Operations

Equations' 5.1, 5.2 and 5.3 are placed in conservative vector form, in order to maintain the accuracy of the solution. The equations become,

$$\frac{\partial Q}{\partial t} + \frac{\partial F}{\partial x} + \frac{\partial G}{\partial y} + \frac{\partial H}{\partial z} = 0 \quad (5.6)$$

$$Q = [\rho, \rho u, \rho v, \rho w, e]^T$$

$$F = [\rho u, \rho u^2 + p - \tau_{xx}, \rho uv - \tau_{xy}, \rho uw - \tau_{xz}, (e + p)u - u\tau_{xx} - v\tau_{xy} - w\tau_{xz} + q_x]^T$$

$$G = [\rho v, \rho vu - \tau_{xy}, \rho v^2 + p - \tau_{yy}, \rho vw - \tau_{yz}, (e + p)v - u\tau_{xy} - v\tau_{yy} - w\tau_{yz} + q_y]^T$$

$$H = [\rho w, \rho wu - \tau_{xz}, \rho wv - \tau_{yz}, \rho w^2 + p - \tau_{zz}, (e + p)w - u\tau_{xz} - v\tau_{yz} - w\tau_{zz} + q_z]^T$$

(u), (v) and (w) are the velocity components. (τ_{ij} , $i, j = x, y, z$) are the viscous stress components. These stresses can be formed from the last two terms in Equation 5.4. (e) is the total energy and (q_i , $i = x, y, z$) are the heat transfer terms. Equation 5.6 is non-dimensionalized, so that the different flows may easily be compared, using parameters, such as Mach, Reynolds and Prandtl Numbers. Solutions from non-dimensionalized equations may also be applied to a flow over different size bodies of the same shape. The equations are transformed from the physical domain into computational domain. This transformation is done, so that the equations may eventually be differenced on a uniformly spaced grid [Reference 22]. Equation 5.6 is further simplified by the *thin layer approximation*, which neglects the viscous terms over the body in the streamwise direction.. The equations can now be cast into finite difference form. This leads to a system of algebraic equations, which are solved implicitly in OVERFLOW.

The system in a very simple form may be written as,

$$[lhs]\{\Delta Q\} = \{rhs\} \quad (5.7)$$

This is called the *delta form* [Reference 4]. The matrix $[lhs]$ and vector $\{rhs\}$ are functions of known flow quantities, such as density, pressure, etc. The vector $\{\Delta Q\}$ is the unknown change in the dependent variables between the current and next time step. For steady flows, when the change in these variables approaches zero, the solution is considered to have *converged* to the best approximation of the original Equations 5.1 through 5.3.

B. OVERFLOW MISSILE FOREBODY MODEL DESCRIPTION

A flow field about a missile forebody was constructed and exercised in OVERFLOW to obtain an acceptable converged solution. Figure 14 shows the missile forebody's surface, but for clarity it does not show the flow grid extending above the surface. The missile is oriented in an X type configuration to the flow. The axis system is body fixed with the same angle of attack definition given in Figure 9. All of the dimensions of the grid have been non-dimensionalized with respect to the missile's maximum diameter. The maximum diameter is one, while the missile's length is approximately 17.5 diameters. The flow grid extends 39 diameters forward of the nose. The grid extends over the missile body's surface at the aft end to 75 diameters and for computational efficiency only one half of the flow field is modeled. The flow is assumed symmetric about the x, z plane. There is no grid in the negative y direction. There is no grid aft of the missile body's rear base.

The grid over the missile forebody's surface is composed of six sub-grids. These define the in-flow over the missile forebody's nose, the body's mid section, the top and bottom canards, and the out-flow over the body's rear section. The sixth grid is an outer flow grid and covers the mid section. These six grids provide 874,125 total grid points to describe the flow field over the missile's forebody.

All these grids overlap each other to varying extents. In the overlapped regions the *chimera* scheme is used to provide interpolation and grid blanking information [Reference 23]. This information is used to allow OVERFLOW to interpolate the flow variables from one grid to the next. The six grids and the interpolation information are provided to the code in two large files called GRID.IN and FORT.2 respectively. The files were from Reference 24. In addition to these files OVERFLOW requires an input file that describes such things as the number of time steps desired, angle of attack, Mach and Reynolds numbers, time step size, boundary conditions, turbulence model, etc. Appendix G lists a typical OVERFLOW input file for the missile forebody. A description of the various input parameters can be found in Reference 20. All OVERFLOW results shown in this thesis were derived from version 1.6ao, 8 February 1994, of the code. The forebody model was initially exercised at a benign flow condition of zero angle of attack, in order to check the flow physics and solution convergence.

C. OVERFLOW MISSILE FOREBODY FLOW SOLUTION CONVERGENCE STUDY

The initial flow field was set for a free stream Mach Number of 0.3, a Reynolds Number of 2.0 million and zero degree angle of attack. Figures 15 through 20 show the plots of the L_2 norm of the vector $\{\Delta Q\}$ in Equation 5.7 at each iteration. The L_2 norm is the square root of the sum of squares of the vector's elements. As the norm approaches zero, the change in the dependent variable also approaches zero, creating a converged solution. The plots in these figures indicate an acceptable converged solution, after 4000 time steps.

D. PMARC MISSILE FOREBODY MODEL DESCRIPTION

A PMARC flow field of the missile forebody was constructed, using 4836 surface panels. As discussed in Chapter IV, wake panels and boundary layers were not included. The geometry of the PMARC missile forebody in Figure 21 was very similar to the OVERFLOW forebody in Figure 14. Some slight differences existed between the canards of the two forebodies. The PMARC model uses the same body fixed coordinate system and angle of attack definition in Figures 9 and 14. Appendix H lists a typical input file for the PMARC missile forebody. A description of the various input parameters can be found in Reference 14. The PMARC model was exercised at various angles of attack and compared to C_p distributions derived from the OVERFLOW forebody model.

Figure 22 compares typical C_p distributions from both the OVERFLOW and PMARC models. This figure indicates that most of the C_p changes occur in the nose region of the missile. This region extends approximately three to five calibers aft the nose, along the x axis. Figure 23 shows the pressure distribution over the first three calibers and the missile body profile. Similar C_p distribution comparisons were made in this region at various angles of attack, in order to demonstrate where the two models begin to differ.

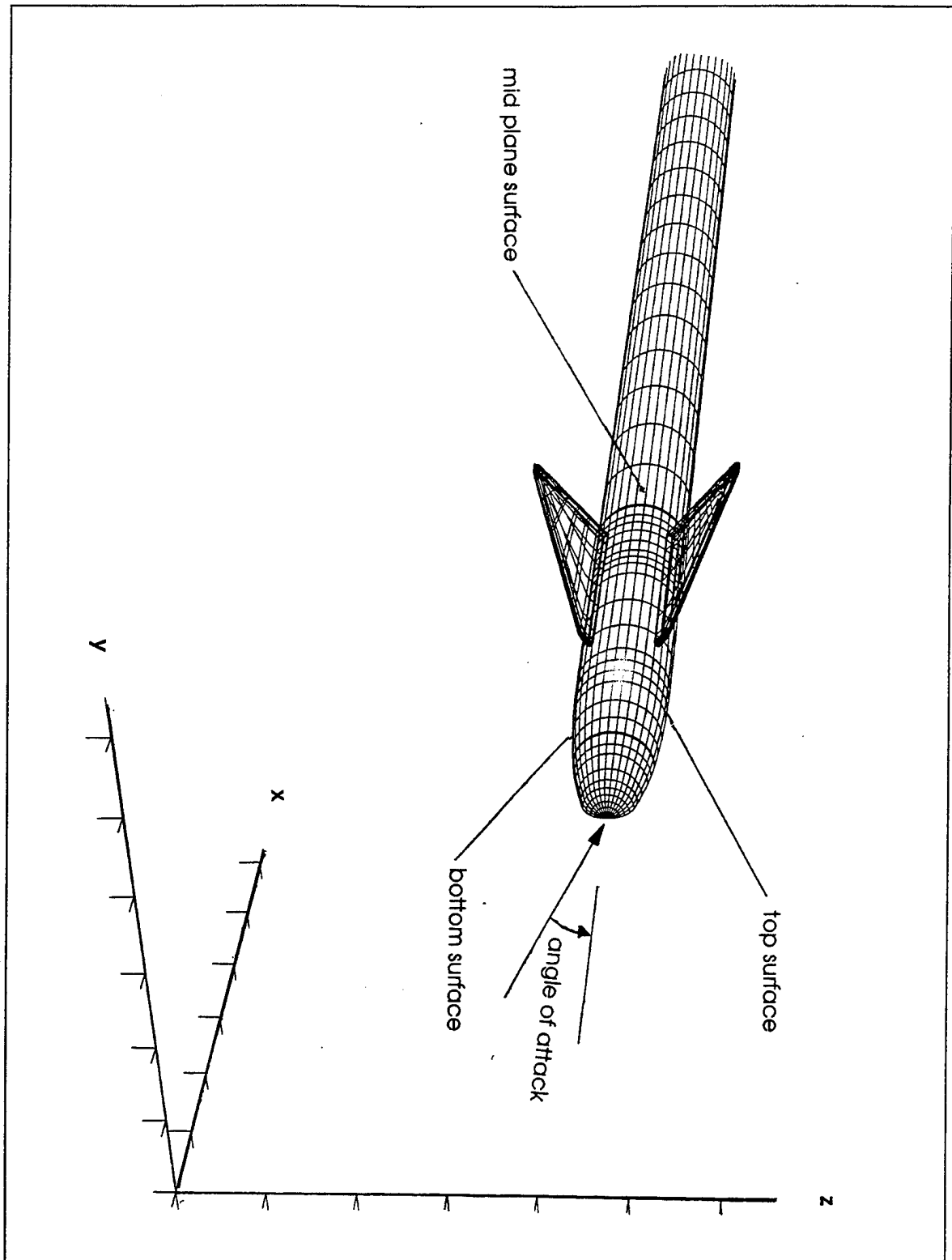


Figure 14. OVERFLOW Missile Forebody Surface.

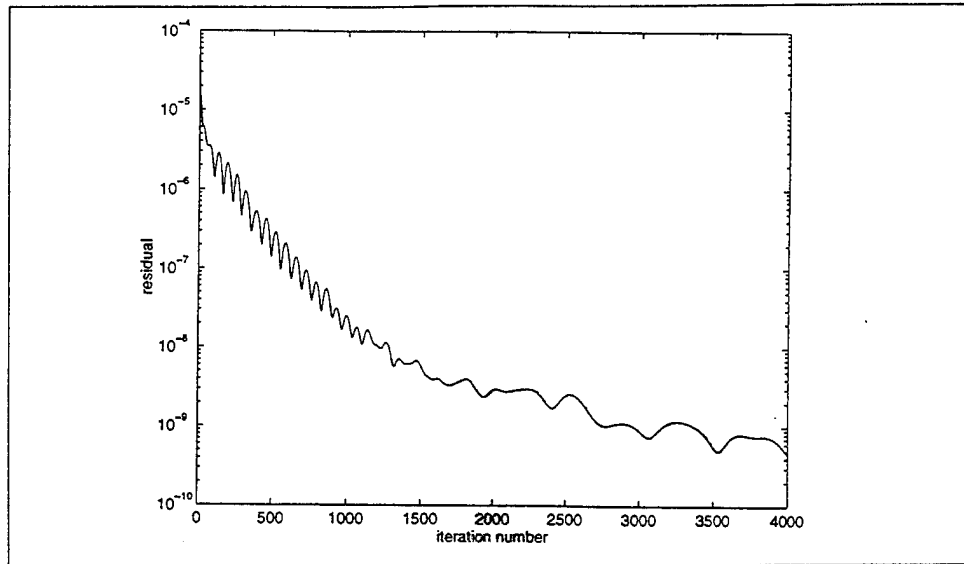


Figure 15. L_2 Norm for the Nose Grid.

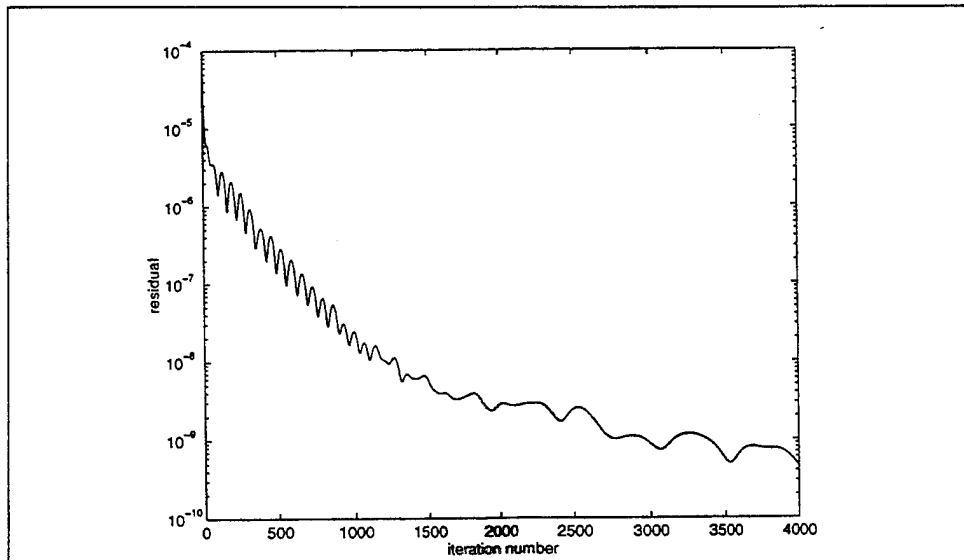


Figure 16. L_2 Norm for the Mid Section Grid.

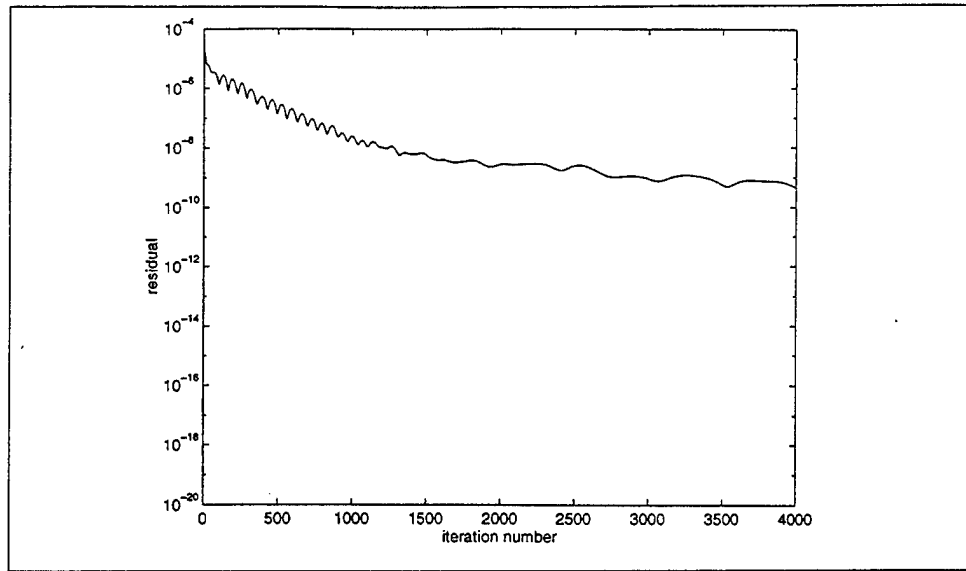


Figure 17. L_2 Norm for the Outer Flow Grid.

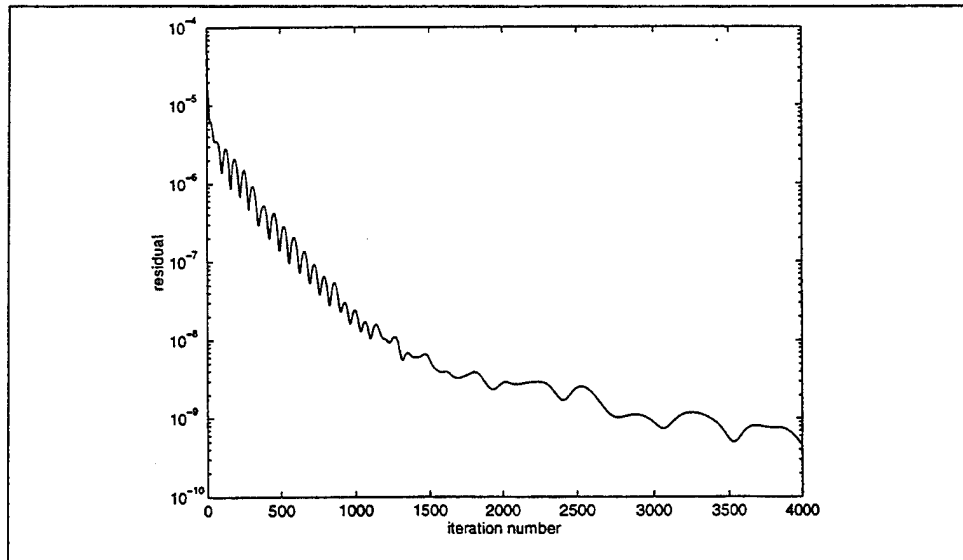


Figure 18. L_2 Norm for the Rear Section Grid.

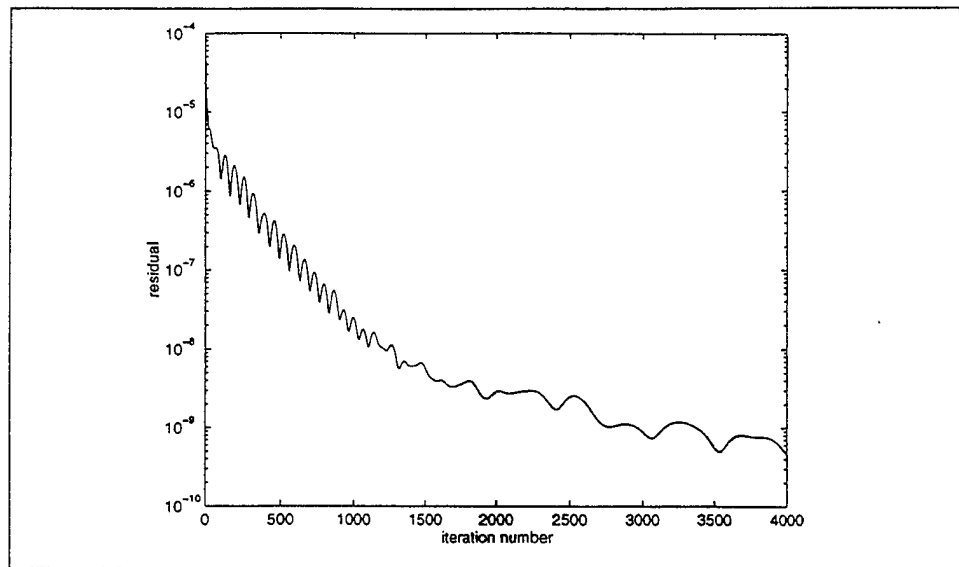


Figure 19. L_2 Norm for the Top Canard Grid.

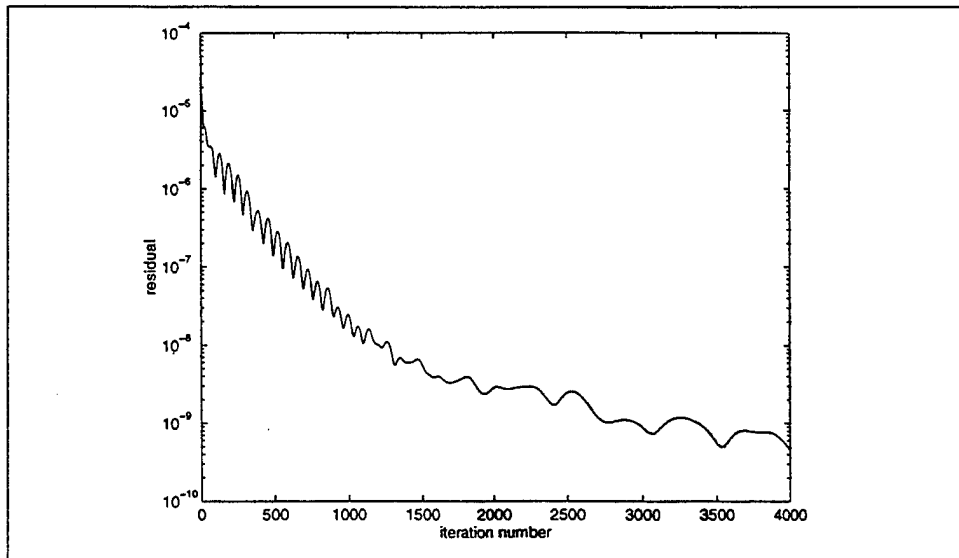


Figure 20. L_2 Norm for the Bottom Canard Grid.

E. COMPUTING TIMES FOR OVERFLOW AND PMARC

Typically OVERFLOW calculations required approximately 10 to 12 cpu days on the Naval Postgraduate School, Cray Y-MP/EL computer for each converged solution. A typical PMARC solution required 1 to 2 cpu days on a Silicon Graphics workstation. The Cray mass storage disk space was accessed across the network for the PMARC runs, since local workstation space was very limited. A PMARC solution could have been obtained much faster, if more local disk space were available.

F. OVERFLOW AND PMARC C_p DISTRIBUTION COMPARISONS AT VARIOUS ANGLES OF ATTACK

Figures 25 through 38 present C_p distributions for the top, mid plane and bottom surfaces of the nose regions of the two missile forebody models. The models were exercised at 0, 2, 6, 10 and 14 degree angles of attack. The scale used in these figures was the same as used in Figure 23. Figures 39 and 53 are enlargements of figures 25 through 38 in the nose region. These figures show how OVERFLOW and PMARC results diverge as the angle of attack increases.

At zero and two degree angles of attack in Figures 24 through 29 and Figures 39 through 44, there are very good agreement between the two models. At six degree angle of attack in Figure 30 through 32 and Figures 45 through 47, some differences begin to appear along the mid plane surface. The differences in the aft portion of the bottom surface are significant in Figure 32.

Figures 33 through 35 and 48 through 50 show C_p distributions, when the models are at 10 degree angle of attack. Both models are in excellent agreement within the first caliber along the x axis. However, Figures 33 through 35 show differences between both models in the aft portion of the nose region.

Figures 36 through 38 and 51 through 53 present the results of the two models at 14 degree angle of attack. These figures show agreement within the first caliber along the x axis, similar to the 10 degree case. Again there are differences between the OVERFLOW and PMARC models at 14 degree angle of attack in the aft portion of the missile's nose.

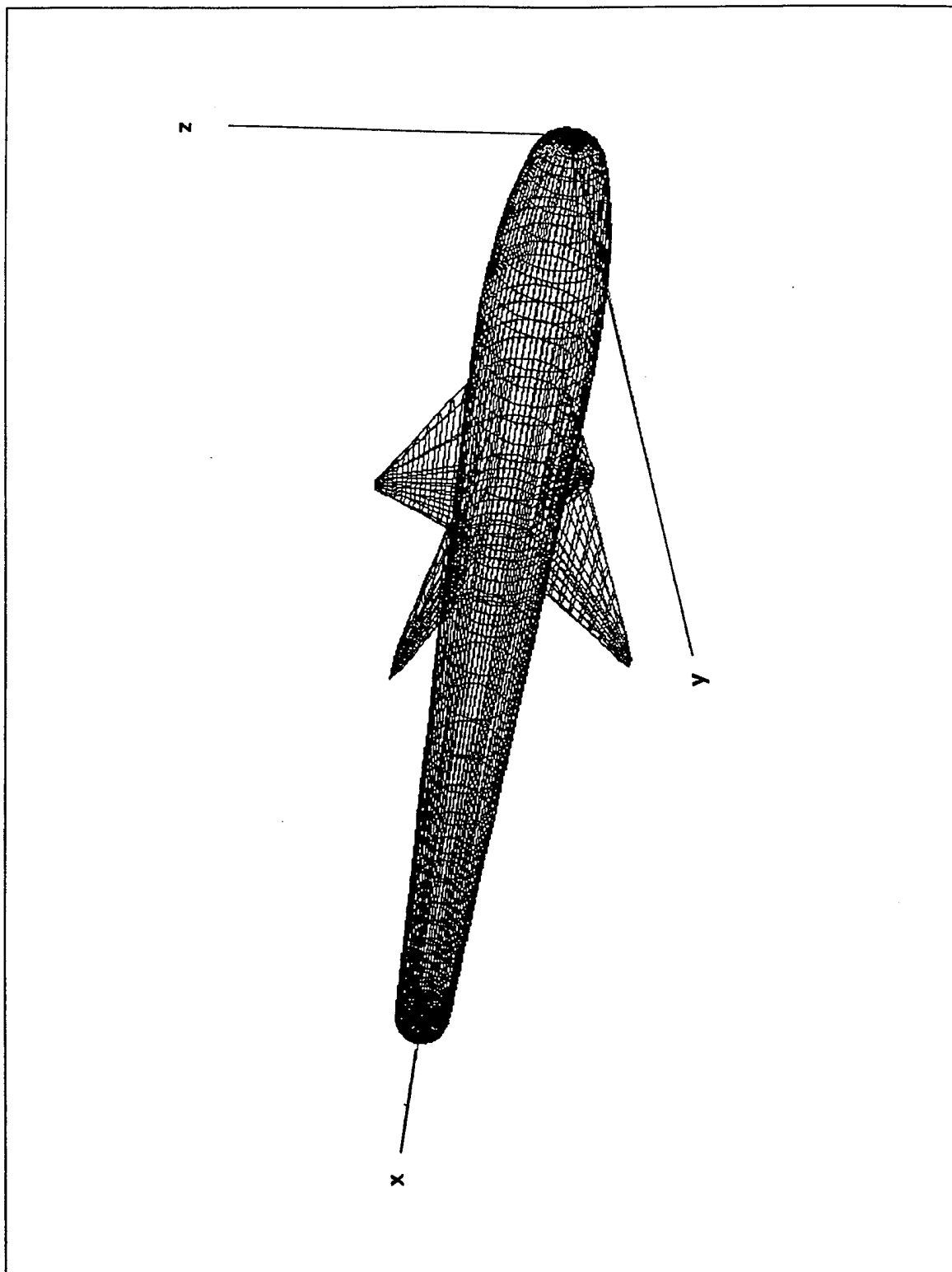


Figure 21. PMARC Missile Forebody Surface.

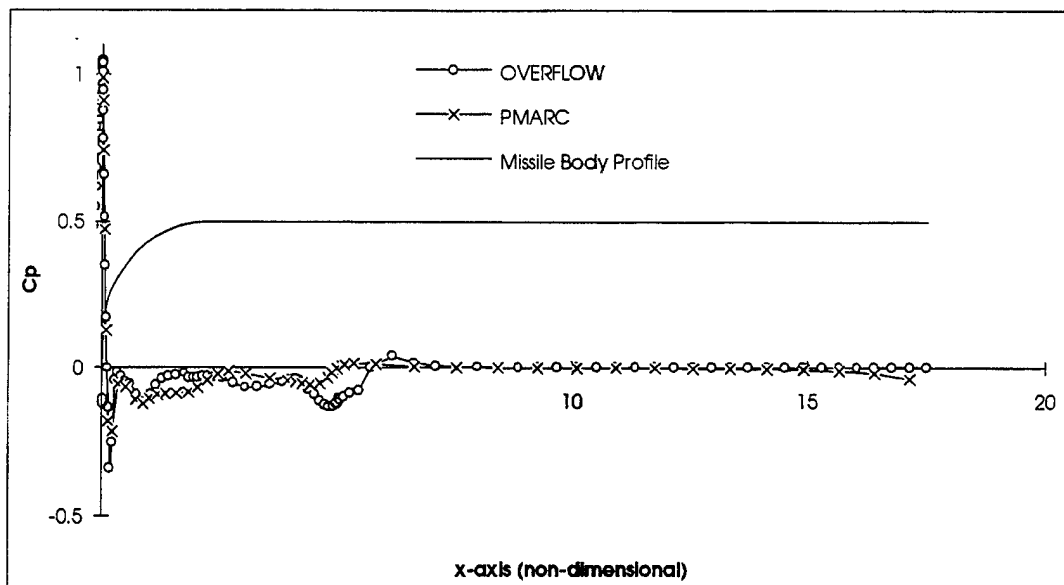


Figure 22. OVERFLOW and PMARC C_p Distributions for the Full Missile Forebody.

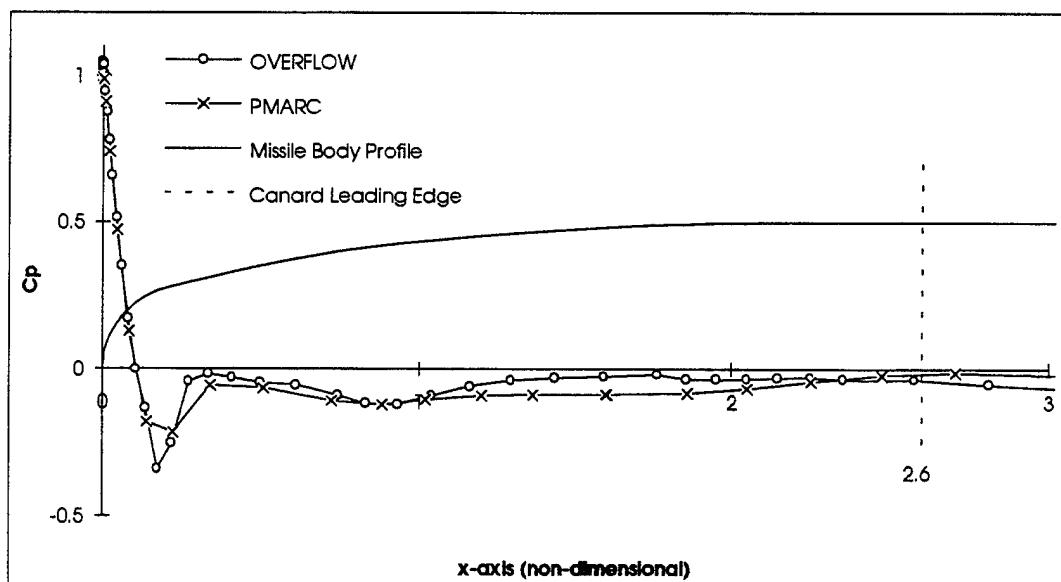


Figure 23. OVERFLOW and PMARC C_p Distributions for the Missile Nose.

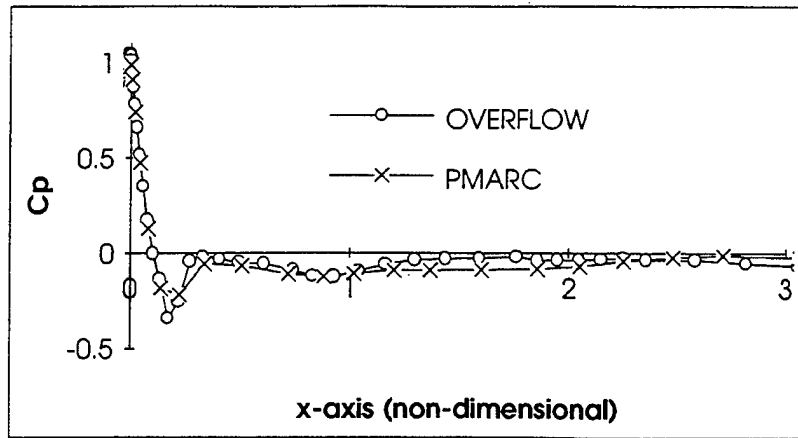


Figure 24. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Zero Degree Angle of Attack on the Top Surface.

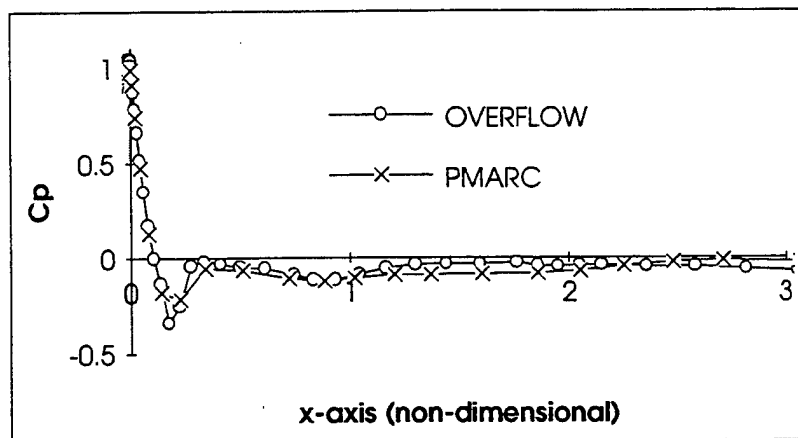


Figure 25. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Zero Degree Angle of Attack on the Mid Plane Surface.

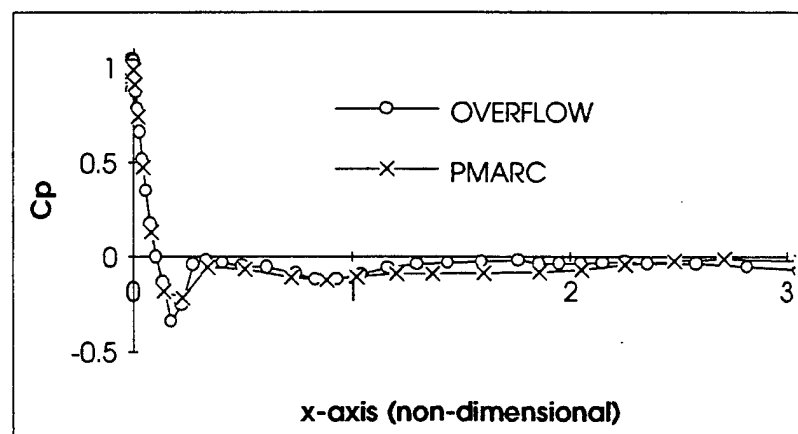


Figure 26. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Zero Degree Angle of Attack on the Bottom Surface.

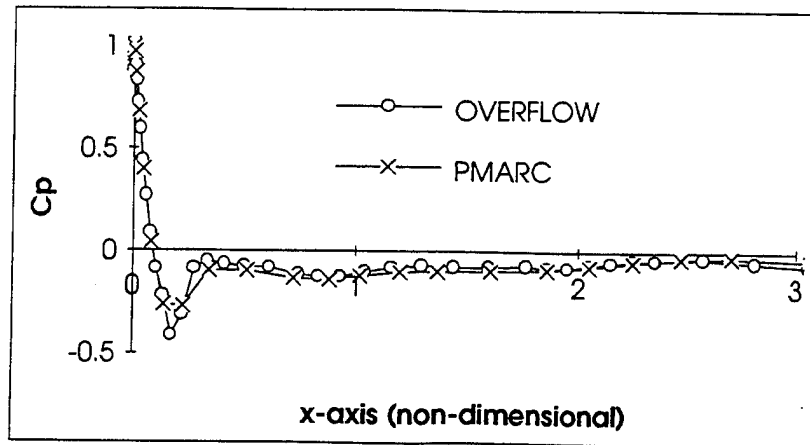


Figure 27. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Two Degree Angle of Attack on the Top Surface.

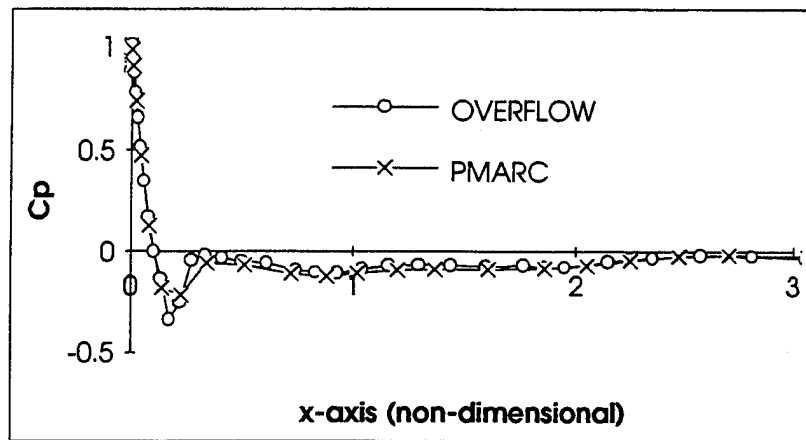


Figure 28. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Two Degree Angle of Attack on the Mid Plane Surface.

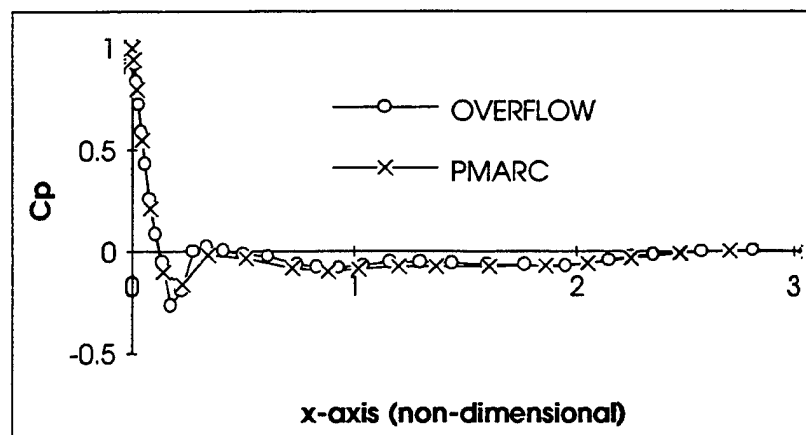


Figure 29. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Two Degree Angle of Attack on the Bottom Surface.

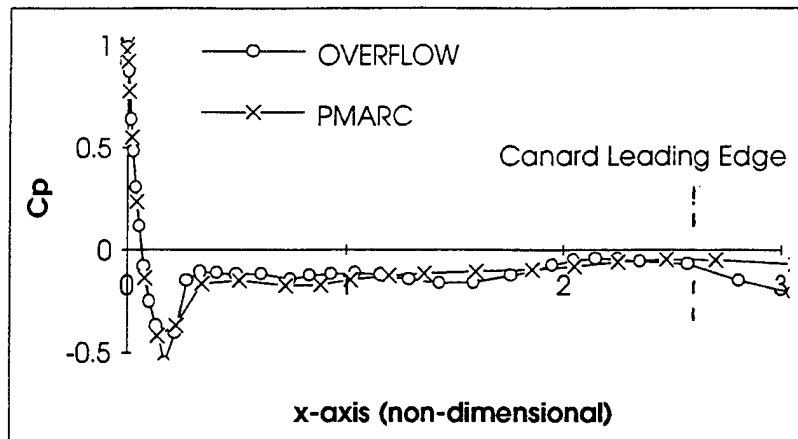


Figure 30. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Six Degree Angle of Attack on the Top Surface.

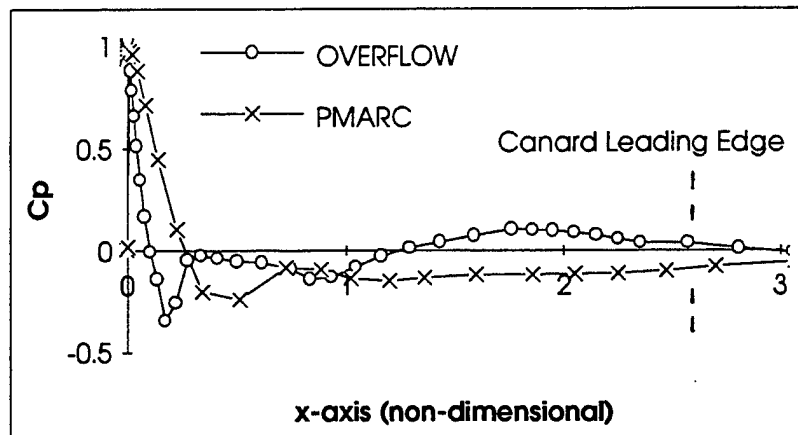


Figure 31. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Six Degree Angle of Attack on the Mid Plane Surface.

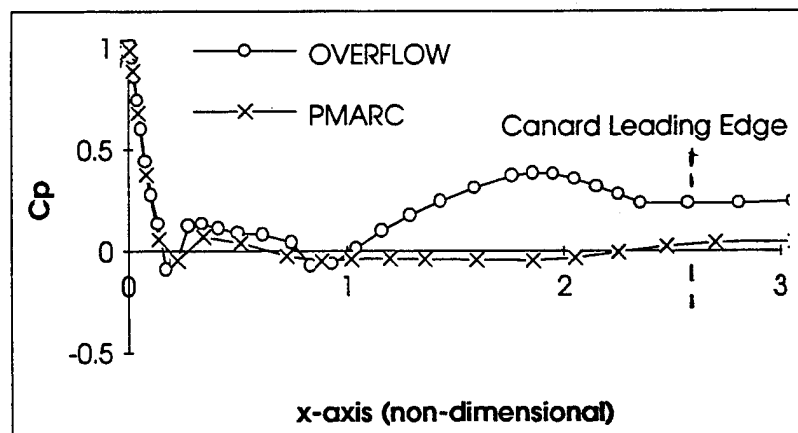


Figure 32. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Six Degree Angle of Attack on the Bottom Surface.

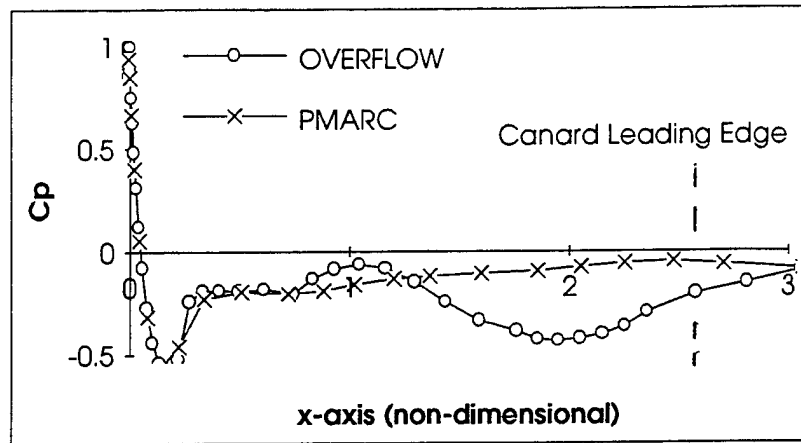


Figure 33. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Ten Degree Angle of Attack on the Top Surface.

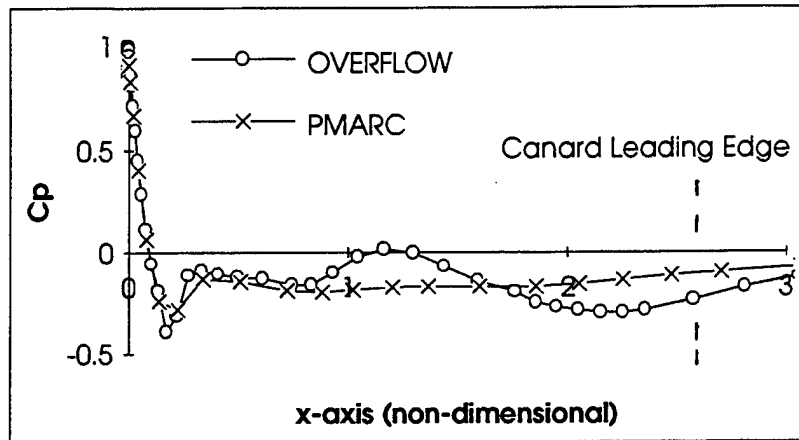


Figure 34. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Ten Degree Angle of Attack on the Mid Plane Surface.

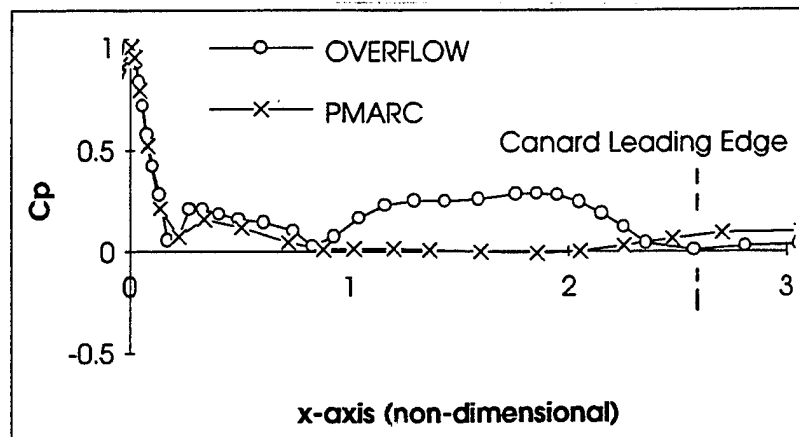


Figure 35. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Ten Degree Angle of Attack on the Bottom Surface.

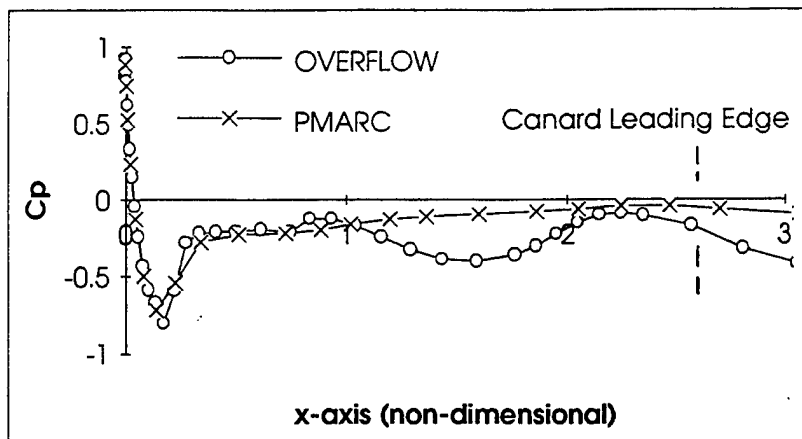


Figure 36. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Fourteen Degree Angle of Attack on the Top Surface.

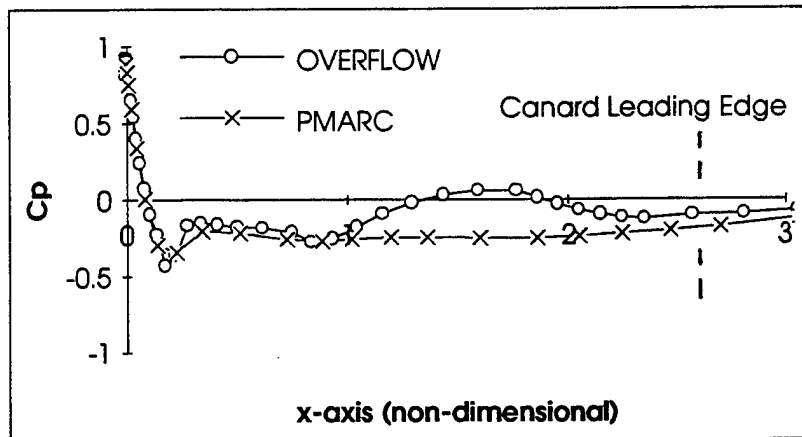


Figure 37. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Fourteen Degree Angle of Attack on the Mid Plane Surface.

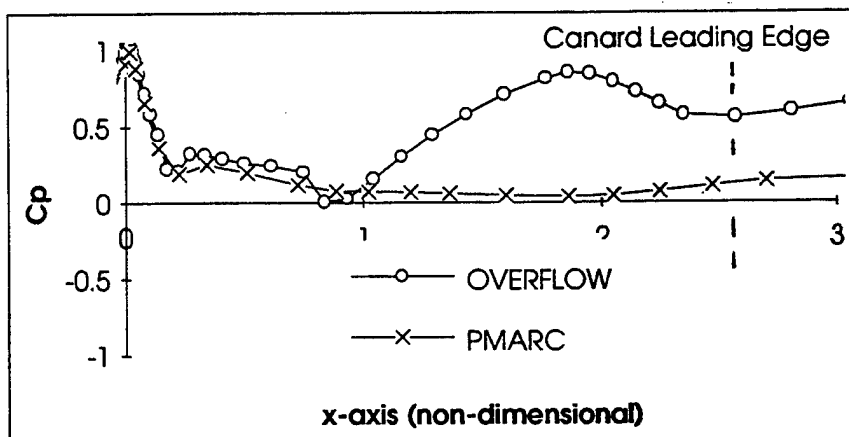


Figure 38. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Fourteen Degree Angle of Attack on the Bottom Surface.

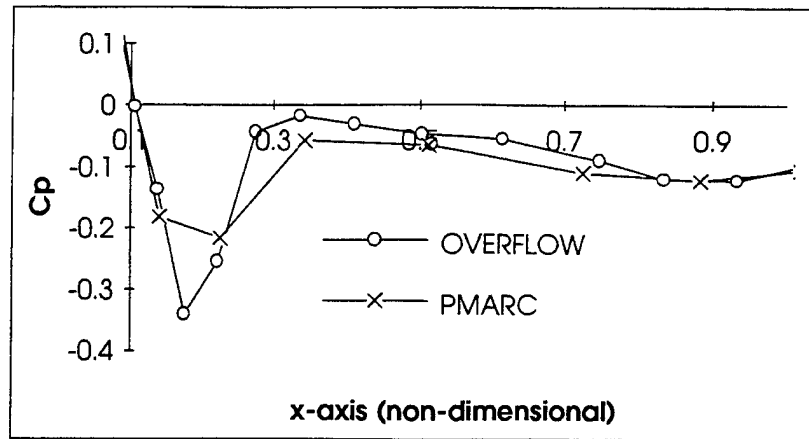


Figure 39. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Zero Degree Angle of Attack on the Top Surface.

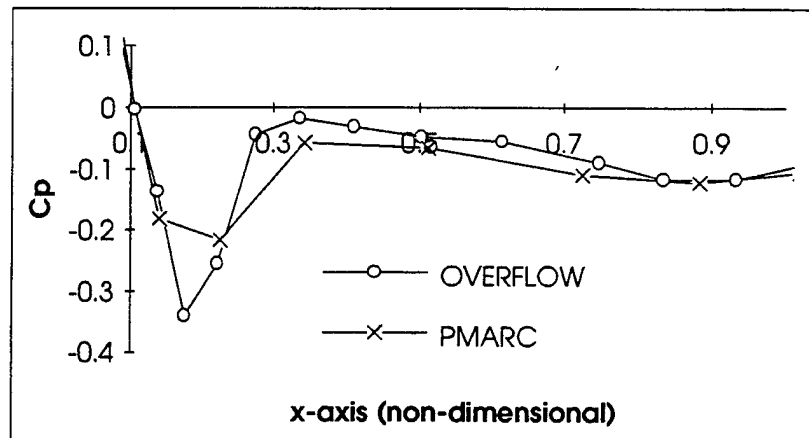


Figure 40. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Zero Degree Angle of Attack on the Mid Plane Surface.

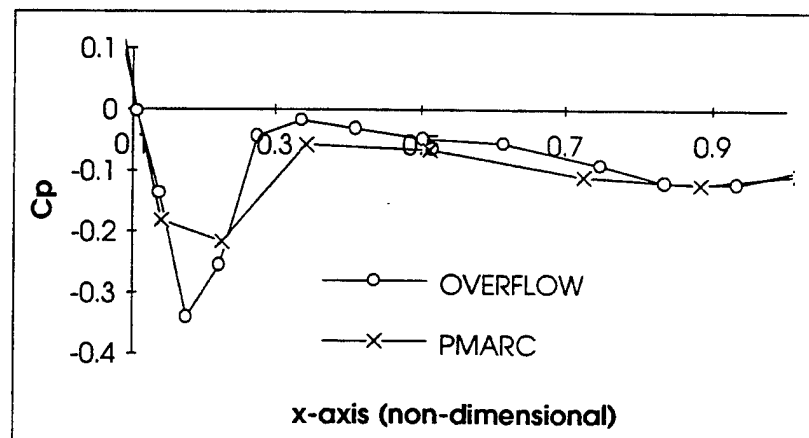


Figure 41. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Zero Degree Angle of Attack on the Bottom Surface.

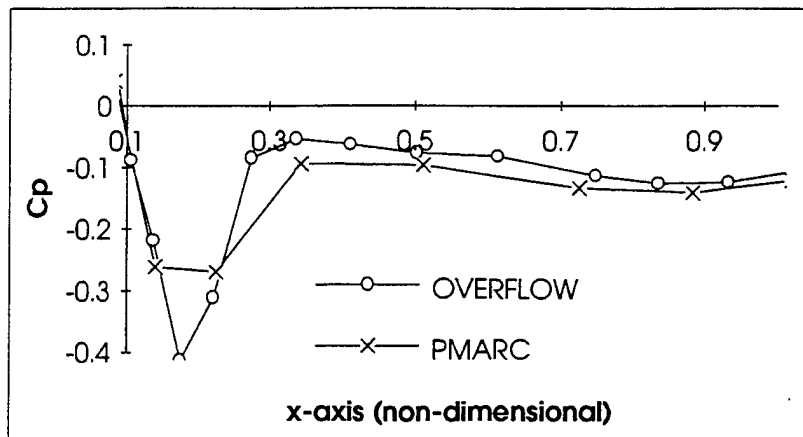


Figure 42. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Two Degree Angle of Attack on the Top Surface.

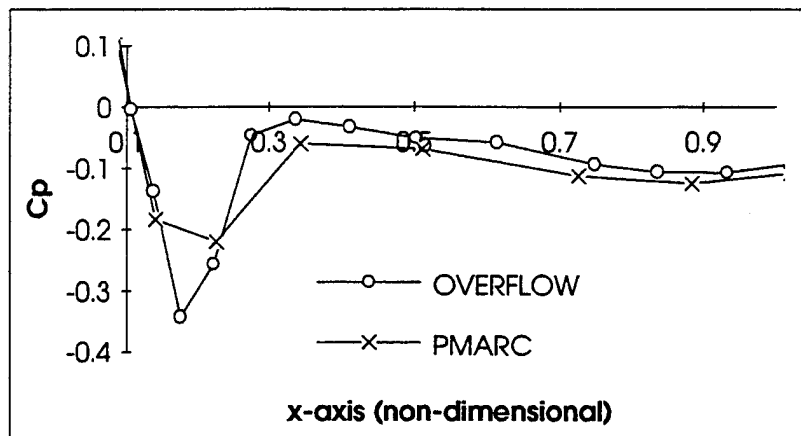


Figure 43. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Two Degree Angle of Attack on the Mid Plane Surface.

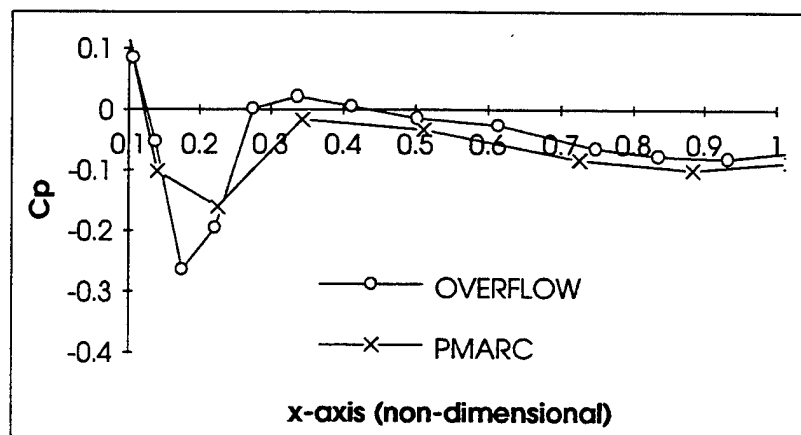


Figure 44. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Two Degree Angle of Attack on the Bottom Surface.

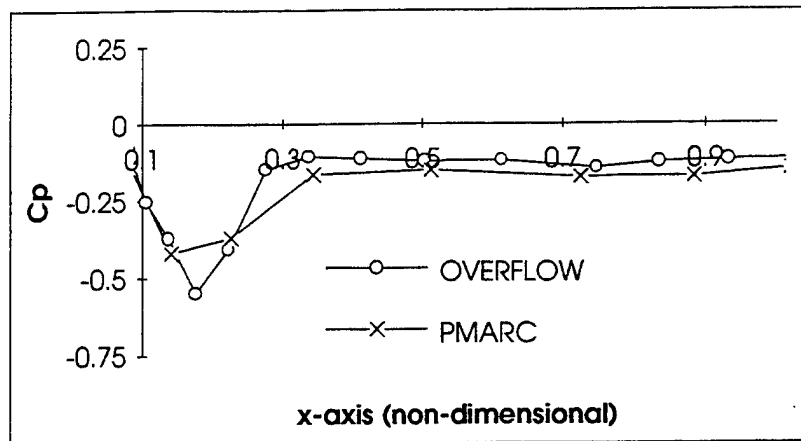


Figure 45. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Six Degree Angle of Attack on the Top Surface.

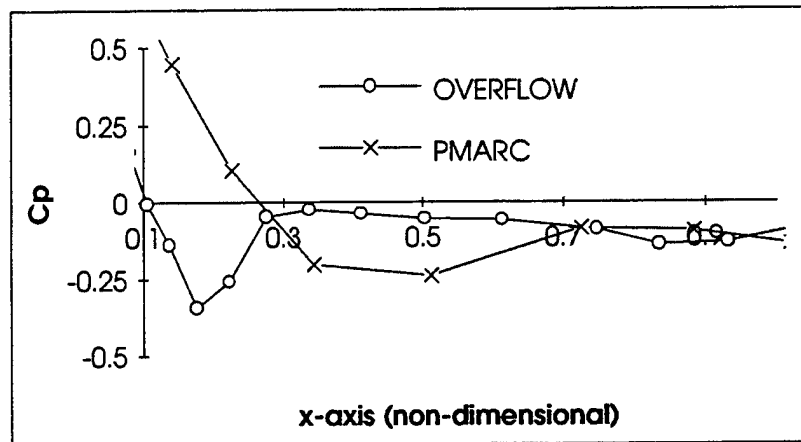


Figure 46. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Six Degree Angle of Attack on the Mid Plane Surface.

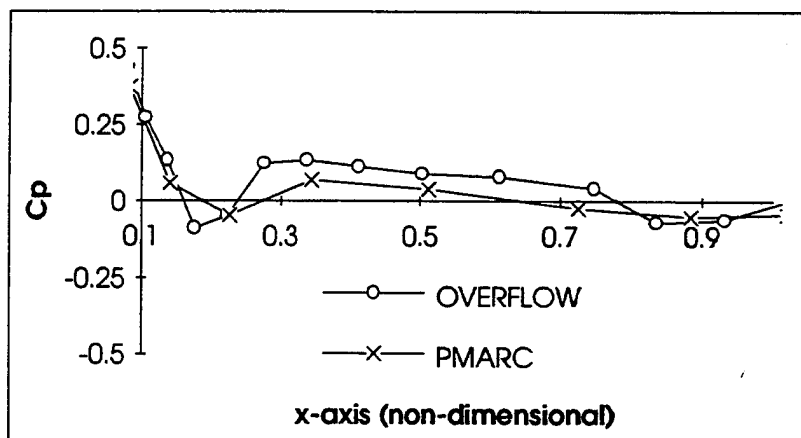


Figure 47. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Six Degree Angle of Attack on the Bottom Surface.

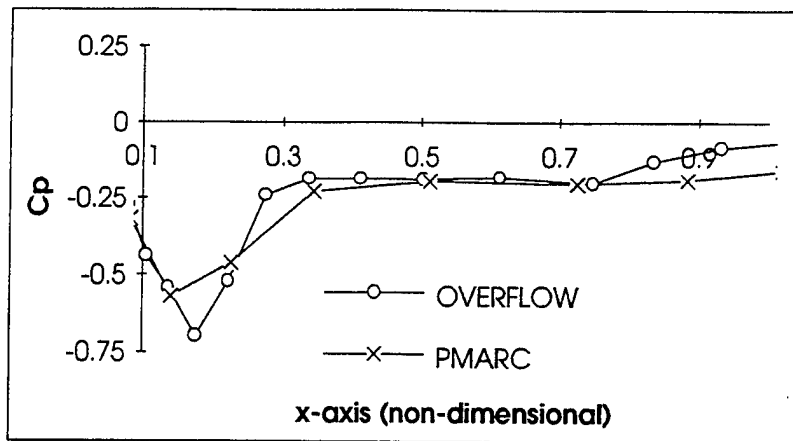


Figure 48. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Ten Degree Angle of Attack on the Top Surface.

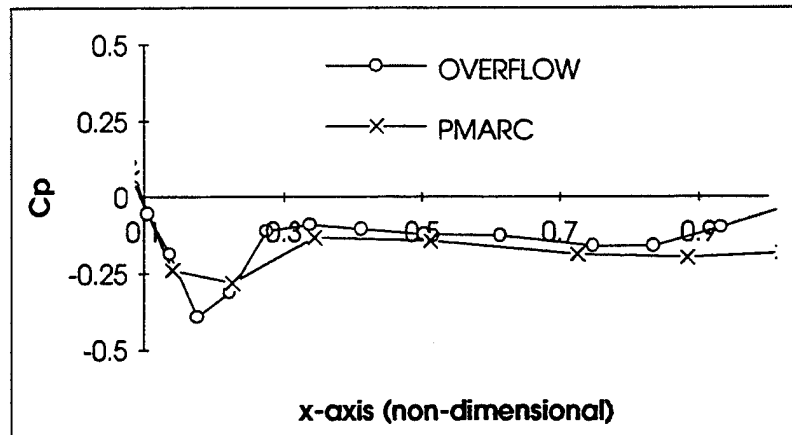


Figure 49. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Ten Degree Angle of Attack on the Mid Plane Surface.

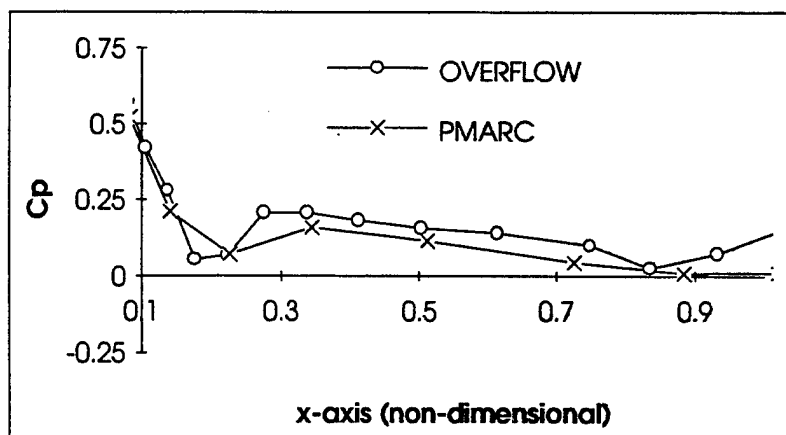


Figure 50. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Ten Degree Angle of Attack on the Bottom Surface.

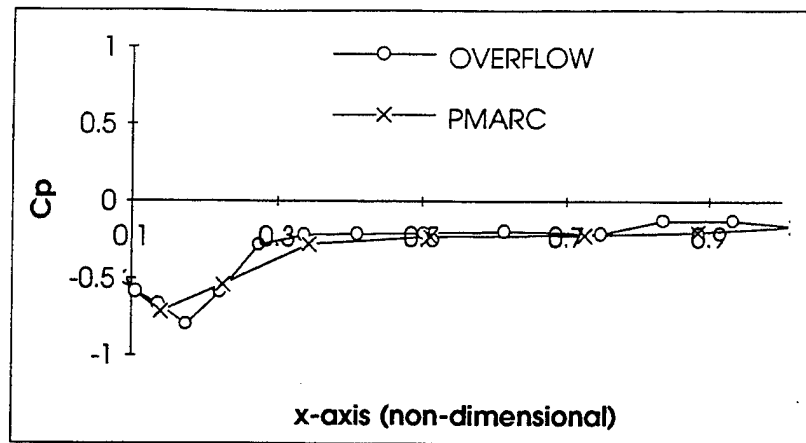


Figure 51. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Fourteen Degree Angle of Attack on the Top Surface.

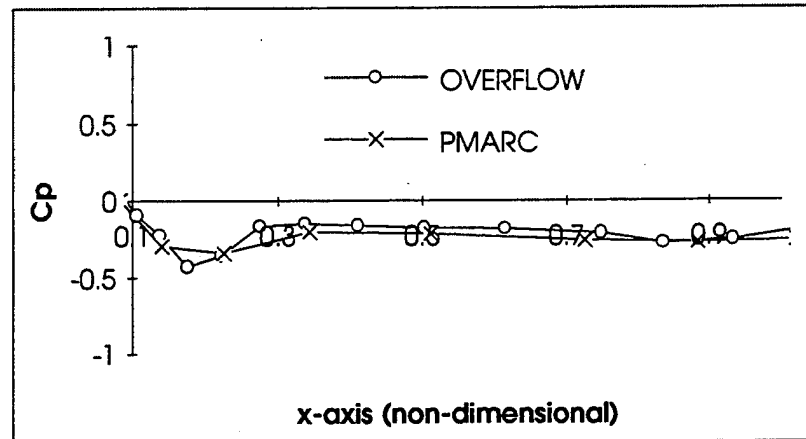


Figure 52. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Fourteen Degree Angle of Attack on the Mid Plane Surface.

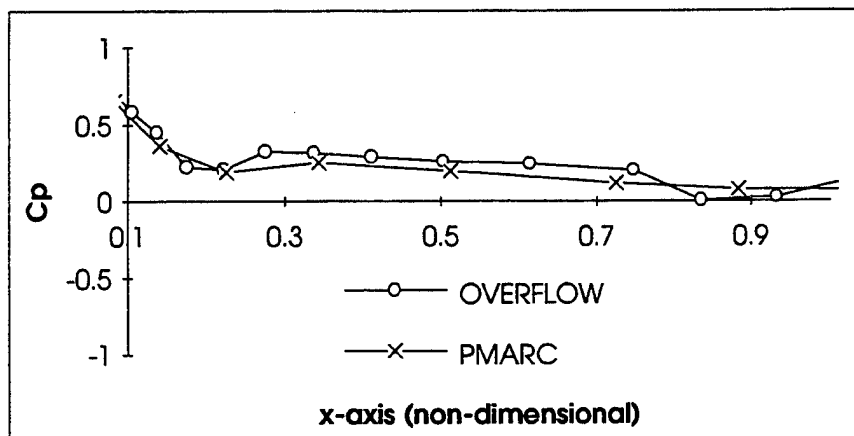


Figure 53. OVERFLOW and PMARC C_p Distributions for the Missile Forebody at Fourteen Degree Angle of Attack on the Bottom Surface.

G. SUMMARY

Missile forebody models were constructed for use in both the OVERFLOW and PMARC Codes. Based on C_p distributions, comparisons were made between these two models for a low Mach number at several angles of attack. Agreement was very good at very low angles of attack. At six degree and higher angles of attack, differences between these two models began to appear.

VI. CONCLUSIONS AND RECOMMENDATIONS

In this thesis the NASA-Ames developed OVERFLOW Navier-Stokes Code was used to obtain solutions for $Mach = 0.3$ and $Reynolds = 2.0e+06$ flow over a typical canard configured missile forebody. In the absence of detailed experimental flow and pressure data, the Navier-Stokes computed pressure distributions were compared with potential flow solutions, i.e., Slender Body Theory, von Karman's Method and the NASA-Ames panel code PMARC. As expected, the first two methods were only of limited applicability, but PMARC provides good agreement for zero and small angles of attack. The PMARC and OVERFLOW computed pressure distributions started to deviate in front and close to the canards. This indicates the presence of flow separation and vortical flow development. Further, work is necessary to display the flow features in more detail, so that the underlying flow physics can be better understood.

APPENDIX A. MAPLE PROCEDURE TO CALCULATE C_p VALUES ON A SPINDLE

```

>
>
> # This MAPLE Procedure calculates the  $C_p$  values about a spindle body of
> # revolution at zero degree angle of attack, using potential
> # theory. The  $C_p$  values are determined in Sections 1 through 3.
>
> # Section 4 determines the non-dimensional longitudinal velocity of
> # Equation 2.10 and compares it to Equation 2.5 to confirm the
> # procedure.
>
> # Sections 5 and 6 investigate the indeterminate behavior of the  $C_p$  or
> # Equation 2.7, as the spindle's radius approaches zero and the  $x$ 
> # coordinate either approaches zero or one respectively.
>


---


>
> # SECTION 1. Calculate non-dimensional longitudinal velocity component,
> # using Equations 2.8, 2.3 and 2.5.
>
> tau:= 0.16;
>
> R:= 2.0*tau* ( xi) - (xi^2) );
>
> R := .320 xi - .320 xi^2
>
> derivative_R:= diff( R, xi );
>
> derivative_R := .320 - .640 xi
>
> eq_2_3:= 2.0 * pi * R * derivative_R;
>
> eq_2_3 := 2.0 pi (.320 xi - .320 xi^2) (.320 - .640 xi)
>
> Integrand_eq_2_5:= ( eq_2_3 * (x-xi) ) / ( ( (x-xi)^2 + r^2 )^(3/2) );
>
> integrand_eq_2_5 := 2.0 * pi (.320 xi - .320 xi^2) (.320 - .640 xi) (x - xi) /
> ( (x - xi)^2 + r^2 )^(3/2)
>
> Eq_2_5:= ( (1) / (4.0*pi) ) * ( int( Integrand_eq_2_5, xi=0.0..1.0 ) );
>
> Eq_2_5 := .2500000000 ( -.2048000000 pi
> ( -21.0 x^2 - 6. x %3 sqrt(%2) + 15.00 x - 3.000 + 6. x^2 %3 sqrt(%2) + 9. x r^2 + 9. x^3 - 3.0 r^2 + %3 sqrt(%2) - 3. r^2 %3 sqrt(%2) )
> / sqrt(%2) + .2048000000 pi
> ( -6. r^2 + 9. x^3 + 6. x^2 %1 sqrt(x^2 + r^2) + 9. x r^2 - 3. r^2 %1 sqrt(x^2 + r^2) - 6. x^2 - 6. x %1 sqrt(x^2 + r^2) + %1 sqrt(x^2 + r^2) ) /
> sqrt(x^2 + r^2) ) / pi
>
> %1 := ln( 2. sqrt(x^2 + r^2) - 2. x )
>
> %2 := x^2 - 2.0 x + 1.00 + r^2
>
> %3 := ln( 2. sqrt(%2) + 2.0 - 2. x )
>


---


>
>
> # SECTION 2. Calculate non-dimensional radial velocity component,
> # using Equation 2.6.

```

```

>
> Integrand_eq_2_6:= ( eq_2_3 ) / ( ( (x-xi)^2 + r^2 )^(3/2) );
                                integrand_eq_2_6:= 2.0  $\frac{\pi (.320 \xi - .320 \xi^2) (.320 - .640 \xi)}{((x - \xi)^2 + r^2)^{3/2}}$ 
> Eq_2_6:= ( (r) / (4.0*pi) ) * ( Int( Integrand_eq_2_6, xi=0.0..1.0 ) );
Eq_2_6:= .2500000000 r ( -.2048000000  $\pi$  ( -4.00 r^2 + 4.0 x^2 + 7.0 x r^2 - 5.0 x^3 - 2. x^2 r^2 + 2. x^4
- 6. x ln( 2.  $\sqrt{\%1}$  + 2.0 - 2. x ) r^2  $\sqrt{\%1}$  - 4. r^4 - 1.0 x + 3. r^2 ln( 2.  $\sqrt{\%1}$  + 2.0 - 2. x )  $\sqrt{\%1}$  ) / ( r^2  $\sqrt{\%1}$  ) +
.2048000000  $\pi$  ( 2. x^4 + r^2 - 2. x^2 r^2 + x^2 + 3. r^2 ln( 2.  $\sqrt{x^2 + r^2}$  - 2. x )  $\sqrt{x^2 + r^2}$ 
- 6. x ln( 2.  $\sqrt{x^2 + r^2}$  - 2. x ) r^2  $\sqrt{x^2 + r^2}$  - 4. r^4 - 3. x r^2 - 3. x^3 ) / ( r^2  $\sqrt{x^2 + r^2}$  ) ) /  $\pi$ 

%1:= x^2 - 2.0 x + 1.00 + r^2
>
>
> # SECTION 3. Calculate the Cp values,
> # using Equation 2.7.
>
> Cp:= ( -2.0*(Eq_2_5) ) - ( (Eq_2_6)**2 ) + ( (Eq_2_5)**2 );
Cp:= -.5000000000 ( -.2048000000  $\pi$ 
( -21.0 x^2 - 6. x  $\sqrt{\%2}$  + 15.00 x - 3.000 + 6. x^2  $\sqrt{\%2}$  + 9. x r^2 + 9. x^3 - 3.0 r^2 +  $\sqrt{\%2}$  - 3. r^2  $\sqrt{\%2}$  )
/  $\sqrt{\%2}$  + .2048000000  $\pi$ 
( -6. r^2 + 9. x^3 + 6. x^2  $\sqrt{x^2 + r^2}$  + 9. x r^2 - 3. r^2  $\sqrt{x^2 + r^2}$  - 6. x^2 - 6. x  $\sqrt{x^2 + r^2}$  +  $\sqrt{x^2 + r^2}$  ) /
 $\sqrt{x^2 + r^2}$  ) /  $\pi$  - .0625000000 r^2 ( -.2048000000
 $\pi$  ( -4.00 r^2 + 4.0 x^2 + 7.0 x r^2 - 5.0 x^3 - 2. x^2 r^2 + 2. x^4 - 6. x  $\sqrt{\%2}$  r^2  $\sqrt{\%2}$  - 4. r^4 - 1.0 x + 3. r^2  $\sqrt{\%2}$  )
r^2  $\sqrt{\%2}$ 
+ .2048000000  $\frac{\pi ( 2. x^4 + r^2 - 2. x^2 r^2 + x^2 + 3. r^2 \sqrt{x^2 + r^2} - 6. x \sqrt{x^2 + r^2} - 4. r^4 - 3. x r^2 - 3. x^3 )^2}{r^2 \sqrt{x^2 + r^2}}$  )^2 /
 $\pi^2$  - .0625000000 ( -.2048000000  $\pi$ 
( -21.0 x^2 - 6. x  $\sqrt{\%2}$  + 15.00 x - 3.000 + 6. x^2  $\sqrt{\%2}$  + 9. x r^2 + 9. x^3 - 3.0 r^2 +  $\sqrt{\%2}$  - 3. r^2  $\sqrt{\%2}$  )
/  $\sqrt{\%2}$  + .2048000000  $\pi$ 
( -6. r^2 + 9. x^3 + 6. x^2  $\sqrt{x^2 + r^2}$  + 9. x r^2 - 3. r^2  $\sqrt{x^2 + r^2}$  - 6. x^2 - 6. x  $\sqrt{x^2 + r^2}$  +  $\sqrt{x^2 + r^2}$  ) /
 $\sqrt{x^2 + r^2}$  )^2 /  $\pi^2$ 

%1:= ln( 2.  $\sqrt{x^2 + r^2}$  - 2. x )

%2:= x^2 - 2.0 x + 1.00 + r^2

%3:= ln( 2.  $\sqrt{\%2}$  + 2.0 - 2. x )
>

```

```

> with(linalg):
Warning: new definition for norm
Warning: new definition for trace

>
> points:= 10;
                                points := 10

> # points:= 48;
>
> increment:= (1.0) / ( points + 1 );
                                increment := .09090909091

>
> X(1):= increment;
                                X(1) := .09090909091

>
> r(1):= 2.0 * tau * ( X(1) - (X(1)^2) );
                                r(1) := .02644628100

> non_dim_ux(1):= evalf( subs( x=X(1), r=r(1), Eq_2_5 ) );
                                non_dim_ux(1) := -.08235051778

> non_dim_ur(1):= evalf( subs( x=X(1), r=r(1), Eq_2_6 ) );
                                non_dim_ur(1) := .2511767847

> cp(1):= evalf( subs( x=X(1), r=r(1), Cp ) );
                                cp(1) := .09482965073

> for i from 2 by 1 to points do
>
> X(i):= X(i-1) + increment;
> r(i):= 2.0*tau* ( X(i) - (X(i)^2) );
> non_dim_ux(i):= evalf( subs( x=X(i), r=r(i), Eq_2_5 ) );
> non_dim_ur(i):= evalf( subs( x=X(i), r=r(i), Eq_2_6 ) );
> cp(i):= evalf( subs( x=X(i), r=r(i), Cp ) );
>
> od:
>
> # writeto( cp );
>
> for i from 0 by 1 to points do
>
> if (i = 0) then
> printf( ' ' ):lprint();
> printf( ' tau = %4.2f, tau ):lprint();
> printf( ' ' ):lprint();
> printf( ' x r ux ur Cp ' ):lprint();
> printf( ' ' ):lprint();
> else
> printf( ' %8.4f %8.4f %8.4f %8.4f %8.4f ', X(i),r(i),non_dim_ux(i),non_dim_ur(i),cp(i)):lprint();
> fi
>
> od;

```


$$\tau = 0.16$$

x	r	ux	ur	Cp
0.0909	0.0264	-0.0824	0.2512	0.0948
0.1818	0.0476	-0.0136	0.1874	-0.0082
0.2727	0.0635	0.0244	0.1300	-0.0663
0.3636	0.0740	0.0458	0.0766	-0.0996
0.4545	0.0793	0.0557	0.0253	-0.1150
0.5455	0.0793	0.0557	-0.0253	-0.1150
0.6364	0.0740	0.0458	-0.0766	-0.0996
0.7273	0.0635	0.0244	-0.1300	-0.0663
0.8182	0.0476	-0.0136	-0.1874	-0.0082
0.9091	0.0264	-0.0824	-0.2512	0.0948

```

>
>
> # SECTION 4. Calculate the approximate non-dimensional, longitudinal velocity component,
> # using Equations 2.10 and compare to Equation 2.5.
>
> Fp1x:= eq_2_3;
                                
$$Fp1x := 2.0 \pi (.320 \xi - .320 \xi^2) (.320 - .640 \xi)$$

> Fp2x:= diff( eq_2_3, xi);
                                
$$Fp2x := 2.0 \pi (.320 - .640 \xi)^2 - 1.2800 \pi (.320 \xi - .320 \xi^2)$$

> Fp3x:= diff( eq_2_3, xi$2);
                                
$$Fp3x := -3.8400 \pi (.320 - .640 \xi)$$

> Fp4x:= diff( eq_2_3, xi$3);
                                
$$Fp4x := 2.4576000 \pi$$

> temp1:= sqrt( xi * (1.0-xi) );
                                
$$temp1 := \sqrt{\xi \sqrt{1.0 - \xi}}$$

> temp2:= (Fp1x) * ((0.5-xi) / (xi*(1.0-xi))) ;
                                
$$temp2 := 2.0 \frac{\pi (.320 \xi - .320 \xi^2) (.320 - .640 \xi) (.5 - \xi)}{\xi (1.0 - \xi)}$$

> temp3:= (Fp2x) * (ln( 2.0*temp1 ) - 1.0);
                                
$$temp3 := (2.0 \pi (.320 - .640 \xi)^2 - 1.2800 \pi (.320 \xi - .320 \xi^2)) (\ln(2.0 \sqrt{\xi \sqrt{1.0 - \xi}}) - 1.0)$$

> temp4:= (Fp3x/ 2.0) * ( 0.5-xi );
                                
$$temp4 := -1.920000000 \pi (.320 - .640 \xi) (.5 - \xi)$$

> temp5:= (Fp4x/48.0) * (1.0 + (4.0*((0.5-xi)**2)));
                                
$$temp5 := .05119999999 \pi (1.0 + 4.0 (.5 - \xi)^2)$$

> Eq_2_10:= (Fp2x/(2.0*pi))*(ln(r))-((1.0/(2.0*pi))*(temp2+temp3+temp4+temp5));
                                
$$Eq_2_10 := .5000000000 \frac{(2.0 \pi (.320 - .640 \xi)^2 - 1.2800 \pi (.320 \xi - .320 \xi^2)) \ln(r)}{\pi} - .5000000000 \left( \frac{2.0 \pi (.320 \xi - .320 \xi^2) (.320 - .640 \xi) (.5 - \xi)}{\xi (1.0 - \xi)} \right)$$


```

$$+ \left(2.0 \pi (.320 - .640 \xi)^2 - 1.2800 \pi (.320 \xi - .320 \xi^2) \right) \left(\ln(2.0 \sqrt{\xi} \sqrt{1.0 - \xi}) - 1.0 \right) \\ - 1.920000000 \pi (.320 - .640 \xi) (.5 - \xi) + .05119999999 \pi (1.0 + 4.0 (.5 - \xi)^2) \Bigg) / \pi$$

```

>
> points:= 10;
                                points := 10

> # points:= 48;
>
> increment:= (1.0) / ( points + 1 );
                                increment := .09090909091

>
> X(1):= increment;
                                X(1) := .09090909091

>
> r(1):=          2.0 * tau * ( (X(1)) - (X(1)^2) );
                                r(1) := .02644628100

> non_dim_ux(1):= evalf( subs( x=X(1), r=r(1), Eq_2_5 ) );
                                non_dim_ux(1) := -.08235051778

> approx_eq_2_5(1):= evalf( subs( xi=X(1), r=r(1), Eq_2_10 ) );
                                approx_eq_2_5(1) := -.0815224471

>
> for i from 2 by 1 to points do
>
>   X(i):=          X(i-1) + increment;
>   r(i):=          2.0*tau* ( (X(i)) - (X(i)^2) );
>   non_dim_ux(i):= evalf( subs( x=X(i), r=r(i), Eq_2_5 ) );
>   approx_eq_2_5(i):= evalf( subs( xi=X(i), r=r(i), Eq_2_10 ) );
>
> od:
>
> # writeto( eq36ux );
>
> for i from 0 by 1 to points do
>
>   if (i = 0) then
>     printf( '          '):lprint();
>     printf( '    tau = %4.2f', tau ):lprint();
>     printf( '          '):lprint();
>     printf( ' x      r      Eq_2_10  Eq_2_5 '):lprint();
>     printf( '          '):lprint();
>   else
>     printf( ' %8.4f  %8.4f  %8.4f  %8.4f', X(i),r(i),approx_eq_2_5(i),non_dim_ux(i)):lprint();
>   fi
>
> od;

tau = 0.16

x          r          Eq_2_10      Eq_2_5

```

0.0909	0.0264	-0.0815	-0.0824
0.1818	0.0476	-0.0141	-0.0136
0.2727	0.0635	0.0222	0.0244
0.3636	0.0740	0.0423	0.0458
0.4545	0.0793	0.0514	0.0557
0.5455	0.0793	0.0514	0.0557
0.6364	0.0740	0.0423	0.0458
0.7273	0.0635	0.0222	0.0244
0.8182	0.0476	-0.0141	-0.0136
0.9091	0.0264	-0.0815	-0.0824

```

>
>
> # SECTION 5. Verify that as r goes to zero and x goes to zero,
> #           Cp values will become indeterminate.
>
> points:= 50;
                                points := 50

>
> increment:= (1.0) / ( 10.0 + 1.0 );
                                increment := .09090909091

> X(1):= increment;
                                X(1) := .09090909091

> r(1):= 2.0 * tau * ( X(1) - (X(1)^2) );
                                r(1) := .02644628100

> cp(1):= evalf( subs( x=X(1), r=r(1), Cp ) );
                                cp(1) := .09482965073

> for i from 2 by 1 to points do
>
>   X(i):= X(i-1) * increment;
>   r(i):= 2.0*tau* ( X(i) - (X(i)^2) );
>   cp(i):= evalf( subs( x=X(i), r=r(i), Cp ) );
>
> od;
> # writeto( x0 );
> for i from 0 by 1 to points do
>
>   if (i = 0) then
>     printf( '          '):lprint();
>     printf( '    tau = %4.2f, tau ):lprint();
>     printf( '          '):lprint();
>     printf( ' x          r          Cp '):lprint();
>     printf( '          '):lprint();
>   else
>     printf( ' %18.16e %18.16e %19.16e ', X(i),r(i),cp(i)):lprint();
>   fi
>
> od;

```

tau = 0.16

x	r	Cp
9.0909090909999990e-02	2.6446280999999999e-02	9.4829650730000020e-02
8.2644628099999978e-03	2.6227716680000002e-03	3.6136853520000001e-01
7.5131480089999985e-04	2.4024010460000001e-04	5.3521760820000008e-01
6.8301345540000010e-05	2.1854937759999998e-05	6.6798813509999999e-01
6.2092132309999992e-06	1.9869358970000001e-06	7.6933566659999997e-01
5.6447393010000001e-07	1.8063155569999999e-07	8.4041487029999995e-01
5.1315811830000005e-08	1.6421058939999999e-08	8.8133818590000002e-01
4.6650738029999991e-09	1.4928236100000000e-09	8.9211484799999996e-01
4.2409761850000004e-10	1.3571123789999998e-10	8.7274545100000001e-01
3.8554328949999998e-11	1.2337385260000000e-11	8.2323001299999998e-01
3.5049389950000002e-12	1.1215804780000001e-12	7.4356853099999998e-01
3.1863081770000003e-13	1.0196186169999998e-13	6.3376100300000016e-01
2.8966437970000004e-14	9.2692601500000018e-15	4.9380743199999994e-01
2.6333125429999999e-15	8.4266001380000004e-16	3.2370781599999998e-01
2.3939204939999999e-16	7.6605455800000010e-17	1.2346215499999999e-01
2.1762913580000001e-17	6.9641323460000009e-18	-1.0692954999999997e-01
1.9784466890000002e-18	6.3310294039999996e-19	-3.6746730099999997e-01
1.7985878989999998e-19	5.7554812759999995e-20	-6.5815109900000002e-01
1.6350799080000001e-20	5.2322557059999992e-21	-9.7898093900000016e-01
1.4864362800000000e-21	4.7565960960000007e-22	-1.3299568230000001e+00
1.3513057090000002e-22	4.3241782680000007e-23	-1.7110787580000002e+00
1.2284597350000001e-23	3.9310711519999992e-24	-2.1223467270000005e+00
1.1167815770000001e-24	3.5737010460000005e-25	-2.5637607600000001e+00
1.0152559790000002e-25	3.2488191319999990e-26	-3.0353208079999998e+00
9.2295998089999999e-27	2.9534719380000003e-27	-3.5370269280000000e+00
8.3905452809999996e-28	2.6849744900000005e-28	-4.0688790870000000e+00
7.6277684370000006e-29	2.4408859000000001e-29	-4.6308772819999993e+00
6.9343349430000001e-30	2.2189871820000004e-30	-5.2230215270000002e+00
6.3039408569999999e-31	2.0172610740000002e-31	-5.8453118119999994e+00
5.7308553250000002e-32	1.8338737040000003e-32	-6.4977481520000007e+00
5.2098684770000003e-33	1.6671579130000000e-33	-7.1803305270000006e+00
4.7362440700000003e-34	1.5155981020000002e-34	-7.8930589520000005e+00
4.3056764270000000e-35	1.3778164570000001e-35	-8.6359334170000004e+00
3.9142512970000000e-36	1.2525604150000002e-36	-9.4089539470000005e+00
3.5584102699999994e-37	1.1386912859999998e-37	-1.0212120480000001e+01
3.2349184269999993e-38	1.0351738970000002e-38	-1.1045433100000000e+01
2.9408349340000003e-39	9.4106717880000007e-40	-1.1908891739999998e+01
2.6734863040000001e-40	8.5551561719999991e-41	-1.2802496430000001e+01
2.4304420949999996e-41	7.7774147040000000e-42	-1.3726247190000000e+01
2.2094928140000002e-42	7.0703770039999999e-43	-1.4680143960000001e+01
2.0086298310000000e-43	6.4276154600000007e-44	-1.5664186760000002e+01
1.8260271189999999e-44	5.8432867799999995e-45	-1.6678375619999997e+01
1.6600246540000000e-45	5.3120788919999998e-46	-1.7722710589999998e+01
1.5091133220000000e-46	4.8291626299999998e-47	-1.8797191539999996e+01
1.3719212019999997e-47	4.3901478459999999e-48	-1.9901818540000004e+01
1.2472010930000000e-48	3.9910434980000007e-49	-2.1036591570000002e+01
1.1338191749999998e-49	3.6282213599999993e-50	-2.2201510679999995e+01
1.0307447049999999e-50	3.2983830560000001e-51	-2.3396575810000005e+01
9.3704064090000006e-52	2.9985300500000003e-52	-2.4621787019999999e+01
8.5185512810000014e-53	2.7259364099999994e-53	-2.5877144220000002e+01

>

>

> # SECTION 6. Verify that as r goes to zero and x goes to one,

```

> #          Cp values will become indeterminate.
>
> X(1):= 0.9500;
                                X(1) := .9500
> r(1):= 2.0 * tau * ( X(1) ) - (X(1)^2 );
                                r(1) := .01520000000
> cp(1):= evalf( subs( x=X(1), r=r(1), Cp ) );
                                cp(1) := .1752243365
> X(2):= 0.999900;
                                X(2) := .999900
> r(2):= 2.0 * tau * ( X(1) ) - (X(1)^2 );
                                r(2) := .01520000000
> cp(2):= evalf( subs( x=X(1), r=r(1), Cp ) );
                                cp(2) := .1752243365
>
> points:= 29;
                                points := 29
>
> increment:= 0.000001;
                                increment := .1 10-5
> for i from 3 by 1 to points do
>
>   X(i):= X(i-1) + increment:
>   r(i):= 2.0*tau* ( X(i) ) - (X(i)^2 ):
>   cp(i):= evalf( subs( x=X(i), r=r(i), Cp ) );
>
> od:
Error, (in evalf/ln) singularity encountered
> # writeto( x1 );
> for i from 0 by 1 to points do
>
>   if (i = 0) then
>     printf( '          '):lprint():
>     printf( '    tau = %4.2f, tau ):lprint():
>     printf( '          '):lprint():
>     printf( ' x          r          Cp '):lprint():
>     printf( '          '):lprint():
>   else
>     printf( ' %18.16f %18.16f %19.16f ', X(i),r(i),cp(i)):lprint():
>   fi
>
> od;

tau = 0.16

x                                r                                Cp
0.94999999999999998  0.01520000000000000  0.17522433650000000

```

0.9998999999999999	0.0152000000000000	0.1752243365000000
0.9999009999999999	0.0000316768640000	0.6476606865000000
0.9999020000000001	0.0000313569280000	0.6553688371999999
0.9999029999999999	0.0000310369920000	0.6422468258999999
0.9999040000000001	0.0000307170560000	0.6393732819000001
0.9999050000000002	0.0000303971200000	0.6585344844000000
0.9999060000000001	0.0000300771840000	0.6561241774000001
0.9999069999999999	0.0000297572480000	0.6536068717000000
0.9999079999999999	0.0000294372800000	0.6398706602999999
0.9999090000000000	0.0000291173440000	0.6489621221999999
0.9999100000000000	0.0000287974080000	0.6580446579000001
0.9999110000000002	0.0000284774720000	0.6431658841000000
0.9999120000000001	0.0000281575360000	0.6400647958000000
0.9999129999999999	0.0000278375680000	0.6507920520000000
0.9999140000000001	0.0000275176320000	0.6343043787000000
0.9999150000000002	0.0000271976960000	0.6448765211000000
0.9999160000000001	0.0000268777280000	0.6425597706000000
0.9999169999999999	0.0000265577920000	0.6392634326000001
0.9999179999999999	0.0000262378560000	0.6357834041000000
0.9999189999999999	0.0000259178880000	0.6482360795000001
0.9999200000000000	0.0000255979520000	0.6293168993000000
0.9999210000000001	0.0000252780160000	0.6416774725000000
0.9999219999999999	0.0000249580480000	0.6547907977000000
0.9999230000000002	0.0000246381120000	0.6354180657000001
0.9999240000000001	0.0000243181440000	0.6494795412000000
0.9999250000000000	0.0000239982080000	0.6623045039000000
0.9999260000000000	0.0000236782400000	0.6253918384000000
0.9999269999999999	0.0000233583040000	-5.8453118119999994

```

>
>
> # file = app_a.txt
>
>
>

```


Page 2

sphere.f

Jul 20 1995 08:29

Page 1

sphere.f

1995 08:29

```

c program sphere
c
c SPHERS calculates the pressure coefficients about a spherical body
c of revolution. It is a slight modification of Chow's implementation of
c vonKarman's procedure for using line shapes along the body's
c symmetric axis to define the body's shape (see Chow, C.R., "An
c Introduction to Computational Fluid Mechanics", Seminar Publishing,
c corrected edition 1983, pp 105 - 117).
c
c This code differs from Chow's implementation in the following ways,
c 1. SPHERS uses double precision. Chow's code is single precision,
c 2. SPHERS loops through different sets of surface points (PI), in order
c to investigate the code's convergence properties, Chow's code does
c not do this.
c 3. SPHERS contains a subroutine to calculate the error norm defined as,
c
c      err = sqrt( sum(((dd)*(q) - (const))*2) )
c
c The number of surface points (m) and the (error) are listed in the
c file "error.out".
c If the (error) grows in magnitude as (m) increases, using double
c precision, then this is an indication that the coefficient
c matrix (dd) is ILL-CONDITIONED.
c
c The file "sphere.out" lists for each set of (m) surface points,
c 1. the calculated coefficient of pressure (cp) and the exact
c 2. the calculated coefficient of pressure (cpx) and the exact
c coefficient of pressure (cpxt) for comparison.
c
c u = non-dimensional free stream velocity, assume u = 1.0.
c a = non-dimensional total sphere radius, assume a = 1.0.
c n = number of surface points (PI), assume n = odd number.
c m = number of (PI) used in each iteration.
c notes: n = m + 1, and n = even number.
c
c parameter( kk=250 )
c real*8 u, a, pi, phi(kk), z(kk), cp(kk), cpx(kk), err
c integer*4 m, n, i
c
c open( unit=20, file='sph.out', status='unknown' )
c open( unit=25, file='err.out', status='unknown' )
c
c open( unit=20, file='sphere.out', status='unknown' )
c open( unit=25, file='error.out', status='unknown' )
c
c Input the non-dimensional free stream velocity (u) and sphere radius (a).
c
c u = 1.0
c a = 1.0
c
c Write headers for output files
c
c write(20,' ' ,
c write(20,' ' , This is the data output file called SPHERE.OUT'
c write(20,' ' , from the program SPHERS.FOR.'
c write(20,' ' ,
c write(20,' ' , ECHO constant input values for the SPHERZ:
c write(20,' ' ,
c write(20,' ' , u (non-dimensional velocity) = ',u
c write(20,' ' , a (non-dimensional length) = ',a
c write(25,' ' ,
c write(25,' ' , This is the data output file called ERROR.OUT.'
c write(25,' ' , from the program SPHERS.FOR.'
c write(25,' ' ,
c write(25,' ' , m, error norm',
c write(25,' ' ,
c
c start looping for different sets of surface points (PI)
c
c do 5, m = 1, 51, 2
c n = m + 1
c
c Calculate the surface points, where,
c (a) is the sphere's radius,
c (xi) is the horizontal axis of the sphere,
c (zi) is the perpendicular distance from (xi).

```



```

c Create right hand side or constant vector (const).
c
do 30, i = 1, (n-1)
  const(i) = { (r(i))**2 } / (2.0)
30 continue
const(n) = 0.0

c Solve for the source strengths (q) vector, using Cramer's Rule.
c
call cramer( dd, const, q, n )

c Calc. the error norm (err) between (const) and (q) vectors
c
call error( n, dd, const, q, err )

c Calc. velocities (v), (vex) and pressure coefficients (cp), (cpex).
c (v) and (cp) are approximated, while (vex) and (cpex) are exact
c values for comparison.
c
do 90, i = 1, (n-1)
  ur = 0.0
  us = 1.0
  do 80, j = 1, n
    ur = ur +
      $      q(j)/r(i)*(s(i)-ss(j+1))/d(i,j+1)-(s(i)-ss(j))/d(i,j))
80 continue
  v(i) = u * sqrt(ur**2 + us**2)
  vex(i) = 1.5 * u * (r(i) / a)
  cp(i) = 1.5 - (v(i)/u)**2
  cpex(i) = 1.0 - (vex(i)/u)**2
90 continue
RETURN
END

c
c
c subroutine cramer( q, a, x, n )
c real*8 determ, denom, cc(kk,kk), a(kk,kk), x(kk)
c integer*4 k, n, i, j

c DETERM is a function called by this subroutine.
c
denom = determ( c, n )
if (denom.eq. 0.0) then
  write(20,*) ' In subroutine CRAMER,'
  write(20,*) ' denom = ' , denom
  write(20,*) ' causes (x(k)) in subroutine to blow up,'
  write(20,*) ' CAN NOT obtain a (q) vector at,'
  write(20,*) ' n = ', (n-1), ' , surface points !!!'
  stop
else
  end if
do 3, k = 1, n
  do 1, i = 1, n
    do 1, j = 1, n
      1 cc(i,j) = c(i,j)
    do 2, i = 1, n
      2 cc(i,k) = a(i)
    3 x(k) = determ( cc, n ) / denom
  RETURN
  END

c
c
c function determ( array, n )
c
c This function calculates the determinate of the matrix (array).
c
parameter ( kk=250 )
real*8 ratio, determ, array(kk,kk), a(kk,kk)
integer*4 i, n, j, m, k, l

```

```

c Note: in this function (a) has been redefined as a dummy index
c counter for the array (a).
c
do 1, i = 1, n
  do 1, j = 1, n
    1 a(i,j) = array(i,j)
  2 k = m + 1
  do 3, i = k, n
    if (a(m,m).eq. 0.0) then
      write(20,*) ' In function DETERM,'
      write(20,*) ' row m is a zero row, the diagonal element = ' ,a(m,m)
      write(20,*) ' causes (ratio) in function to blow up,'
      write(20,*) ' CAN NOT obtain a (q) vector at,'
      write(20,*) ' m = ', (n-1), ' , surface points !!!'
      stop
    else
      ratio = a(i,m)/a(m,m)
      do 3, j = k, n
        3 a(i,j) = a(i,j) - a(m,j)*ratio
      if (a(m,m).eq. n-i) go to 4
      m = m+1
      go to 2
    4 determ = 1.0
    do 5, i = 1, n
      5 determ = determ * a(i,i)
    RETURN
    END

c
c
c subroutine error( n, dd, const, q, err )
c parameter ( kk=250 )
c real*8 dd(kk,kk), const(kk), q(kk), temp1(kk), temp2(kk), sum, err
c integer*4 i, j, n
c do 1, m = 1, n
  do 1, i = 1, n
    temp1(i) = (dd(i,j) * q(j)) + temp1(i)
  end do
  do 1, i = 1, n
    temp2(i) = const(i) - temp1(i)
  end do
  sum = 0.0
  do 1, i = 1, n
    sum = sum + (temp2(i))**2
  end do
  err = sqrt( sum )
  RETURN
  END

c
c

```

Page 2

Jul 20 1995 08:39 sph-out

Jul 20 1995 08:39 sph.out

```

This is the data output file called SPHERE.OUT
from the program SPHERE.FOR.

ECHO constant input values for the SPHERE

u (non-dimensional velocity) = 1.0000000000000000
a (non-dimensional length) = 1.0000000000000000

FOR m (surface points) = 1

zi, xi

6.123031769111866E-017 1.0000000000000000

cp, cpx

-2.2.134953560336429E-001 -1.2500000000000000

FOR m (surface points) = 3

zi, xi

7.071067811865475E-001 7.071067811865476E-001
6.123031769111866E-017 1.0000000000000000
7.071067811865476E-001 7.071067811865475E-001

cp, cpx

-1.1.257221742451583E-001 -1.2500000000000000
-1.248285350965735 -1.2500000000000000
-1.1.257221742451621E-001 -1.2499999999999998E-001

FOR m (surface points) = 5

zi, xi

8.660254037844385E-001 5.0000000000000000
4.9999999999999998E-001 8.660254037844387E-001
1.722556201536435E-016 1.0000000000000000
5.0000000000000001E-001 8.660254037844386E-001
8.660254037844387E-001 4.9999999999999999E-001

cp, cpx

4.381030903084913E-001 4.3749999999999993E-001
-6.81281927776808E-001 -6.8760000000000001E-001
-1.249402855261340 -1.2500000000000000
-6.88125327778910E-001 -6.8750000000000001E-001
4.381030903084913E-001 4.3750000000000002E-001

FOR m (surface points) = 7

zi, xi

9.238795325112867E-001 9.238795325112867E-001
7.071067811865475E-001 7.071067811865476E-001
3.826834323365089E-001 3.826834323365089E-001
9.238795325112867E-001 9.238795325112867E-001
6.704951288346657E-001 6.704951288346657E-001
7.071067811865475E-001 7.071067811865476E-001
3.826834323365089E-001 3.826834323365089E-001

cp, cpx

6.704951288346657E-001 6.704951288346657E-001
-1.2500000000000000E-001 -1.2500000000000000E-001
-9.204951288346656E-001 -9.204951288346656E-001
-1.2500000000000000E-001 -1.2500000000000000E-001
-9.204951288346656E-001 -9.204951288346656E-001
7.071067811865475E-001 7.071067811865476E-001
3.826834323365089E-001 3.826834323365089E-001

FOR m (surface points) = 9

zi, xi

```

Jul 20 1995 08:39 sph.out

[illegible]

[illegible]

8. 819212643843551E-001	cp,	9. 66666941248600052E-001	cpex	4. 7139673602259378E-001
9. 23879531212887E-001		9. 9665767025471725E-001		3. 26634347235440238E-001
9. 55694033571210887E-001		9. 78493703060169E-001		2. 907864727255440238E-001
9. 9078528040033030E-001		2. 6147878440068977E-001		1. 95090323201612878E-001
9. 95194782667219695E-001		2. 6147878440068977E-001		9. 8017140323956060E-002
		4. 7958551180460252E-002		
		4. 7958551180460252E-002		
		1. 7328151180493639E-001		
		3. 91151562665681359E-001		
		6. 605523239796580E-001		
		7. 9277231212866798E-001		
		9. 4046682564974319E-001		
		-1. 0973152121706834E-001		
		-1. 0973152121706834E-001		
		-1. 12727264578133108E-001		
		-1. 25722675037948280E-001		
		-1. 087554500978258E-001		
		-1. 187554500978258E-001		
		9. 362165781183331E-001		
		7. 63206993303669E-001		
		5. 66534491490021E-001		
		3. 54280130001932E-001		
		-1. 321005999999999E-001		
		-1. 321005999999999E-001		
		5. 45668013120910E-002		
		4. 97153601591631E-001		
		6. 6661472087100575E-001		
				9. 783834440753631E-001
				9. 783834440753631E-001
				8. 1400331288348657E-001
				6. 704951288348657E-001
				5. 000105121470521E-001
				3. 055166812684147E-002
				9. 4476612684147E-002
				1. 250000000000000E-001
				-3. 4476612684147E-002
				-5. 555186811407489E-001
				-1. 250000000000000E-001
				-9. 20434062013380336E-001
				-1. 16436447407536E-001
				-1. 16436447407536E-001
				-1. 2283834407536E-001
				-1. 250000000000000E-001
				-1. 2283834407536E-001
				-1. 16436447407536E-001
				-1. 060403331380336E-001
				-9. 20434062013380336E-001
				-5. 555186811407489E-001
				-3. 4476612684147E-002
				-1. 250000000000000E-001
				-9. 20434062013380336E-001
				-1. 16436447407536E-001
				-1. 2283834407536E-001
				-1. 250000000000000E-001
				-1. 2283834407536E-001
				-1. 16436447407536E-001
				-1. 060403331380336E-001
				-9. 20434062013380336E-001
				-5. 555186811407489E-001
				-3. 4476612684147E-002
				-1. 250000000000000E-001
				-9. 20434062013380336E-001
				-1. 16436447407536E-001
				-1. 2283834407536E-001
				-1. 250000000000000E-001
				-1. 2283834407536E-001
				-1. 16436447407536E-001
				-1. 060403331380336E-001
				-9. 20434062013380336E-001
				-5. 555186811407489E-001
				-3. 4476612684147E-002
				-1. 250000000000000E-001
				-9. 20434062013380336E-001
				-1. 16436447407536E-001
				-1. 2283834407536E-001
				-1. 250000000000000E-001
				-1. 228383440753

Jul 20 1995 08:39		sph.out		Page 11	
CPX		CPX		CPX	
9.961946980917456E-001	9.715574274765017E-002	9.829087221367340E-001	9.321541983841467E-001	9.829087221367340E-001	9.321541983841467E-001
-44.071105461353730	9.321541983841467E-001	9.492785792574932E-001	9.492785792574932E-001	9.492785792574932E-001	9.492785792574932E-001
-25.3025299183895340	9.492785792574932E-001	9.367999985088500E-001	9.367999985088500E-001	9.367999985088500E-001	9.367999985088500E-001
-20.201826827455240	9.367999985088500E-001	5.981360600973565E-001	5.981360600973565E-001	5.981360600973565E-001	5.981360600973565E-001
-15.38916487962320	5.981360600973565E-001	2.59726612413767E-001	2.59726612413767E-001	2.59726612413767E-001	2.59726612413767E-001
-11.48116366865850	2.59726612413767E-001	7.03541988725624E-001	7.03541988725624E-001	7.03541988725624E-001	7.03541988725624E-001
-6.433637442917979	7.03541988725624E-001	3.23541988725624E-001	3.23541988725624E-001	3.23541988725624E-001	3.23541988725624E-001
-4.45940872673186	3.23541988725624E-001	-5.07726612413767E-001	-5.07726612413767E-001	-5.07726612413767E-001	-5.07726612413767E-001
-3.23541988725624E-001	-5.07726612413767E-001	-6.87500000000000E-001	-6.87500000000000E-001	-6.87500000000000E-001	-6.87500000000000E-001
-1.463243376840265	-6.87500000000000E-001	-8.401360600973571E-001	-8.401360600973571E-001	-8.401360600973571E-001	-8.401360600973571E-001
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-1.271581302215210	-9.867999985088505E-001	-1.099778579257494	-1.099778579257494	-1.099778579257494	-1.099778579257494
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Page 12

out

Jul 20 1995 08:39

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6.190133724672086E-001	6.190133724672086E-001	6.190133724672086E-001	6.190133724672086E-001
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2.815504534307124E-001	2.815504534307124E-001	2.815504534307124E-001	2.815504534307124E-001
-9.3989043563559E-002	-9.3989043563559E-002	-9.3989043563559E-002	-9.3989043563559E-002
-2.815504534307124E-001	-2.815504534307124E-001	-2.815504534307124E-001	-2.815504534307124E-001
-4.574905428138412E-001	-4.574905428138412E-001	-4.574905428138412E-001	-4.574905428138412E-001
-6.190133724672086E-001	-6.190133724672086E-001	-6.190133724672086E-001	-6.190133724672086E-001
-7.613081770410183E-001	-7.613081770410183E-001	-7.613081770410183E-001	-7.613081770410183E-001
-8.766815187239193E-001	-8.766815187239193E-001	-8.766815187239193E-001	-8.766815187239193E-001
-9.865844930066699E-001	-9.865844930066699E-001	-9.865844930066699E-001	-9.865844930066699E-001
-2.150826345526386	-2.150826345526386	-2.150826345526386	-2.150826345526386
7.165774289855862E-001	7.165774289855862E-001	7.165774289855862E-001	7.165774289855862E-001
9.139462433814665E-001	9.139462433814665E-001	9.139462433814665E-001	9.139462433814665E-001
8.766815187239193E-001	8.766815187239193E-001	8.766815187239193E-001	8.766815187239193E-001
7.613081770410183E-001	7.613081770410183E-001	7.613081770410183E-001	7.613081770410183E-001
6.190133724672086E-001	6.190133724672086E-001	6.190133724672086E-001	6.190133724672086E-001
4.574905428138412E-001	4.574905428138412E-001	4.574905428138412E-001	4.574905428138412E-001
2.815504534307124E-001	2.815504534307124E-001	2.815504534307124E-001	2.815504534307124E-001
-9.3989043563559E-002	-9.3989043563559E-002	-9.3989043563559E-002	-9.3989043563559E-002
-2.815504534307124E-001	-2.815504534307124E-001	-2.815504534307124E-001	-2.815504534307124E-001
-4.574905428138412E-001	-4.574905428138412E-001	-4.574905428138412E-001	-4.574905428138412E-001
-6.190133724672086E-001	-6.190133724672086E-001	-6.190133724672086E-001	-6.190133724672086E-001
-7.613081770410183E-001	-7.613081770410183E-001	-7.613081770410183E-001	-7.613081770410183E-001
-8.766815187239193E-001	-8.766815187239193E-001	-8.766815187239193E-001	-8.766815187239193E-001
-9.865844930066699E-001	-9.865844930066699E-001	-9.865844930066699E-001	-9.865844930066699E-001

```
5.224985647159489E-001  8.526401643540922E-001
5.877852522924731E-001  8.090169943749475E-001
6.494480483301837E-001  7.604059656000309E-001
7.071067811865476E-001  7.071067811865475E-001
7.604059656000309E-001  6.494480483301837E-001
8.090169943749475E-001  5.877852522924731E-001
8.526401643540922E-001  5.224985647159488E-001
8.910065241883679E-001  4.539904997395468E-001
9.238795325112867E-001  3.826834323650898E-001
9.510565162951535E-001  3.090169943749474E-001
9.723699203976766E-001  2.334453638559054E-001
9.876883405951378E-001  1.564344650402309E-001
9.96917337331280E-001  7.845909572784494E-002
```

cp,

cpex

```
8.805393639067629E-001  9.861493831695299E-001
9.027353434792926E-001  9.449385808320477E-001
8.511179947685362E-001  8.773823397119137E-001
7.664369495328547E-001  7.851441186718157E-001
6.564562148195853E-001  6.704951288348657E-001
5.255108328774930E-001  5.362584088290321E-001
3.774260578885954E-001  3.857393122069899E-001
2.161384311068086E-001  2.226441186718156E-001
4.580818259550300E-002  5.098877317025950E-002
-1.292247151860526E-001  -1.250000000000000E-001
-3.045329477985960E-001  -3.009887731702601E-001
-4.757068227851423E-001  -4.726441186718159E-001
-6.384599405293646E-001  -6.357393122069905E-001
-7.887313331453566E-001  -7.862584088290325E-001
-9.227818362308864E-001  -9.204951288348656E-001
-1.037282781301297  -1.035144118671816
-1.129395086758662  -1.127382339711914
-1.196836872040474  -1.194938580832048
-1.237937891815081  -1.236149383169530
-1.251678365058649  -1.250000000000000
-1.237718124165071  -1.236149383169530
-1.196394403699731  -1.194938580832048
-1.128723005893080  -1.127382339711913
-1.036368004554203  -1.035144118671816
-9.216015679375731E-001  -9.204951288348656E-001
-7.872478831464250E-001  -7.862584088290320E-001
-6.366135399787071E-001  -6.357393122069900E-001
-4.734060719099040E-001  -4.726441186718159E-001
-3.016426483135478E-001  -3.009887731702596E-001
-1.255511579397279E-001  -1.249999999999999E-001
5.053388664749303E-002  5.098877317025972E-002
2.22780643714663E-001  2.226441186718158E-001
3.854537560597832E-001  3.857393122069903E-001
5.360442244551401E-001  5.362584088290323E-001
6.703425006687579E-001  6.704951288348660E-001
7.850426542218033E-001  7.851441186718159E-001
8.773211116376738E-001  8.773823397119139E-001
9.449068531375339E-001  9.449385808320477E-001
9.861318536673343E-001  9.861493831695299E-001
```

```
In subroutine CRAMER,
denom= 0.000000000000000E+000
causes (x(k)) in subroutine to blow up,
CAN NOT obtain a (q) vector at,
n= 41, surface points !!!
```


APPENDIX D. ERR.OUT FILE FROM THE SPHERE CODE

Jul 19 1995 05:25		err.out	Page 1
This is the data output file called ERROR.OUT from the program SPHERE.FOR.			
number of surface points		error norm	
1		1.130377350882890E-016	
3		4.042327637553116E-015	
5		5.884010837789714E-014	
7		3.335424922697721E-013	
9		7.049768158406932E-013	
11		1.204559719731763E-011	
13		2.989187292091445E-010	
15		2.420649654610353E-009	
17		1.827890796692023E-007	
19		2.427334109556196E-007	
21		4.866208540023427E-006	
23		2.494981699304525E-006	
25		3.456497292250762E-005	
27		2.347015589325810E-003	
29		1.638605492134842E-002	
31		1.014824650891806E-001	
33		6.804778254335419E-001	
35		6.117182181739560	
37		2.740841302405531E-001	
39		1.655699712754890E-002	

APPENDIX E. PMARC INPUT FILE FOR A SPHERE

[illegible]

[illegible][illegible]

APPENDIX F. PMARC INPUT FILE FOR A SPINDLE

[illegible]

JUL 27 1995 08:22

DATAS

Page 1

```

SPIHOLE BODY (AOA=0, PANELS=144)
4BIN2  LPUTP= 1, LSTOUT= 1, LSTRFQ= 0, LENRUN= 0,
4BIN3  LSTWAB= 1, LSTWAK= 0, LSTCPW= 0,
4BIN4  MAXC= 0, SOLARC= 0.0005,
4BIN5  HNTSEP= 0, DSTSEP= 0.5,
4BIN6  RSTM= 1.0, RGRF= 0.0, REF= 5.0, RCORES= 0.05
4BIN7  VINF= 0.05,
4BIN8  VINF= 335.19, VSOUND= 1117.3,
4BIN9  ALDEG= 0.0, YAWDEG= 0.0, PHIDOT= 0.0, THEDOT= 0.0,
4BIN10 PSIDOT= 0.0,
4BIN11 THEMAX= 0.0, PSIMAX= 0.0,
4BIN12 WRX= 0.0, WRY= 0.0,
4BIN13 DTHAX= 0.0, DTHAY= 0.0,
4BIN14 DTHX= 0.0, DTHY= 0.0,
4BIN15 RPHY= 0.0, RPHY= 0.7854, SSPAN= 1.00,
4BIN16 RMPX= 0.0, RMPY= 0.00,
4BIN17 HBCSGE= 0, HBCSGE= 0, HCOZNE= 0,
4BIN18 C2DUB= 0.0, VREF= 0.0,
4BIN19 NORL= 0,
4BIN20 NOCL= 0,
4BIN21 KXAN= 0.0, KSTDE= 0, NEMHAB= 0, NEMSID= 0,
4BIN22 HBLT= 0,
4BIN23 ASEMX= 0.00, ASEMY= 0.00,
4BIN24 ASCAL= 1.00, ATHTET= 0.00,
4BIN25 APXX= 0.00, APYY= 0.00,
4BIN26 APZZ= 0.00,
4BIN27 AHXX= 0.00, AHYY= 1.00,
4BIN28 AHZZ= 0.00,
4BIN29 CONFX= 0.0000, CONFY= 0.0000,
4BIN30 CONFX= 0.0000, CONFY= 0.0000,
4BIN31 CHXZ= 0.0000, CHYZ= 0.0000,
4BIN32 CPXX= 0.0000, CPYY= 0.0000,
4BIN33 CHZZ= 0.0000,
4BIN34 IRV= 0.0, IDPAT= 2, MAKE= 0, KCOMP= 1, KASS= 1, IPATSYM= 0,
4BIN35 IPATCH= 0,
4BIN36 STX= 0.000000E+00, STY= 0.0, STZ= 0.0, SCALE= 0.0,
4BIN37 ALF= 0.0, THETA= 0.0,
4BIN38 TWR= 0,
4BIN39 TINTS= 3,
4BIN40 TINTS= 0,
4BIN41 TINTS= 0,
4BIN42 TINTS= 0,
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4BIN45 TINTS= 0,
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4BIN57 TINTS= 0,
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4BIN276 TINTS= 0,
4BIN277 TINTS= 0,
4BIN278 TINTS= 0
```


APPENDIX G. OVERFLOW INPUT FILE FOR A MISSILE FOREBODY

Jun 9 1995 11:30 aoa03_1.in Page 2

[illegible]

Jun 9 1995 11:30 aoa031.in Page 1

```

$GLOBAL
NETPFS= 1000, RESTRF=.F., MEAVE = 50, CHIRPA=-.Z., MQT = 0, $END
$TJOIN
ALPHA = 0.0, FSMACH= 0.1000, REY= 2.0E+06, TINF= 520.000, $END
$VARGM $END

$GDONAM
NAME = 'HEMISPHERE CYL. NOSE ( GE_1, 23x9x71 )', $END
SNITERS $END
SNETPRN
$FINACU
IRHS = 1, ILHS = 1, IDI88 = 1, $END
DT = 0.20, ITIME= 1, CFLMIN=2, CFLMAX=200, TTOSO = 1.00, $END
$SNOACU
ISPEC = 2, DIS2 = 2.00, DIS4 = 0.02,
SHOO = 1.000, EFSE = 0.060, $END
$VIENP
VISCJ = .F., VISCX = -.T., VISCL = -.T.,
VTORG = 1,
NTURS = 1,
ITVP = 1,
ITDIR = 3,
ITDIR = 2,
JTLS = 23,
KTLS = 93,
LTLS = 93,
MTLS = 1,
LTLS = 71,
TLPAR1 = 0.9, $END
$BCINP
NBC = 5, 12, 12, 5, 40,
IBTTP = 15, 12, 12, 5, 40,
IDIR = 1, 1, -2, 1, -2,
JPCS = 1, 23, 23, 23, 23,
KPCS = 1, 1, 93, 1, 1,
RPCS = 93, 93, 93, 93, 93,
LCES = 1, 1, 1, 1, 1, 71,
LECS = 71, 71, 71, 1, 71,
$END
$SCENTP $END

$GDONAM
NAME = 'Bot. Middle Cy1, Mis. surface ( GE_2, 39x3x49 )', $END
$NITERA $END
$NETPRN
$FINACU
IRHS = 1, ILHS = 1, IDI88 = 1, $END
DT = 0.20, ITIME= 1, CFLMIN=0, CFLMAX=200, TTOSO = 1.00, $END
$SNOACU
ISPEC = 2, DIS2 = 2.00, DIS4 = 0.02,
SHOO = 1.000, EFSE = 0.060, $END
$VIENP
VISCJ = .F., VISCX = .T., VISCL = -.T.,
VTORG = 1,
NTURS = 1,
ITVP = 1,
ITDIR = 3,
ITDIR = 1,
JTLS = 39,
KTLS = 1,
MTLS = 93,
LTLS = 41,
TLPAR1 = 0.9,
$END
$BCINP
NBC = 3, 12, 12, 5,
IBTTP = 12, 12, 12, 5,
IDIR = 2, -2, 3,
JPCS = 1, 1, 31,
KPCS = 1, 93, 31,
RPCS = 1, 93, 93,
LCES = 1, 93, 93,
LECS = 1, 1, 1, 1, 1,

```


\$SCHEINP \$END

\$ORDNAM

NAME = 'Lower Canard Body (gr_5, 51x94x33)', \$END

\$NITERS \$END

\$METPRM

IRRS = 0, ILRS = 2, IDIRS = 2, \$END

\$TIMACU

DT = 0.04, ITIME = 1, CFLMIN = 0, CFLMAX = 200, TPOSO = 1.00, \$END

\$SMOACU

ISPEC = 2, DIS2 = 2.00, DIS4 = 0.02,

SMOO = 1.000, EPSE = 0.060, \$END

\$VISIMP

VISCJ = .T., VISCK = .T., VISCL = .T.,

NTURB = 2,

ITITP = 1, 1,

ITDIR = 3, 1,

JTILS = 1, 1,

JTLE = 51, 51,

KTILS = 1, 1,

KTLE = 94, 94,

LTILS = 1, 1,

LTLE = 33, 33,

TLPAR1 = 0.9, 0.9,

\$END

\$BCINP

NBC = 4,

IBTTP = 15, 10, 5, 5,

IBDIR = -1, 2, 3, 1,

JBCE = 51, 1, 1, 1,

JBCE = 51, 51, 51, 1,

KBCE = 1, 1, 1, 1,

KBCE = 94, 1, 94, 94,

LBCE = 1, 1, 1, 1,

LBCE = 33, 33, 1, 33,

\$END

\$SCHEINP \$END

\$ORDNAM

NAME = 'Upper Canard Body (gr_6, 51x94x33)', \$END

\$NITERS \$END

\$METPRM

IRRS = 0, ILRS = 2, IDIRS = 2, \$END

\$TIMACU

DT = 0.04, ITIME = 1, CFLMIN = 0, CFLMAX = 200, TPOSO = 1.00, \$END

\$SMOACU

ISPEC = 2, DIS2 = 2.00, DIS4 = 0.02,

SMOO = 1.000, EPSE = 0.060, \$END

\$VISIMP

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APPENDIX H. PMARC INPUT FILE FOR A MISSILE FOREBODY

Aug 8 1995 09:51

DATA5

Page 2

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Page 7

DATA5

Aug 8 1995 09:51 DATA5 Page 8

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